

TEMPORARY REVISION

AMM-TR-OÄM 42-369

GMU 44B Magnetomer

This Temporary Revision AMM-TR-OÄM 42-369 is approved in conjunction with the Optional Design Change Advisory OÄM 42-369 and is valid in conjunction with the Airplane Maintenance Manual (AMM) until this Temporary Revision has been incorporated into the AMM.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the AMM.

The technical information contained in this document has been approved under the authority of DOA No. EASA.21J.052.

Doc. No.	Section	Affected Pages
7.02.15	31-40-00	13a, 14a, 220a, 221a, 222a, 223a
	92-00-00	2a, 2aa

Instruction

- Print this document double-sided on yellow paper (odd pages and "a" pages on front sides, even pages and "aa" pages on reverse sides).
- Insert this cover page as the first page of the AMM.
- Insert the other pages of this Temporary Revision in front of the corresponding AMM pages.

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AFFECTED CHAPTERS:

CHAPTER 31
INDICATING SYSTEMS
Section 31-40
Central Computers

2. Description

The following paragraphs are amended to read:

F. GRS 77/79 Attitude, Heading and Reference Unit (AHRs)

The GRS 77/79 provides airplane attitude and related flight data to the cockpit displays. The GRS 77/79 receives data from the air data computer, the GMU 44/B magnetometer and GPS signals from the GIA 6X W. The GRS communicates with both the GIA 6X Ws and the cockpit displays via ARINC 429 digital interface. The GRS 77/79 is located next to the avionics rack in the rear fuselage, just aft of the rear baggage frame.

I. GMU 44/B Magnetometer

The magnetometer senses magnetic field information. Data is sent to the GRS 77/79 ARHS for processing. The magnetometer receives power from the GRS 77/79 and communicates with the GRS 77/79 using RS-485 digital interface. The magnetometer is located in the right outer wing and can be accessed through a panel in the lower surface of the wing.

MAINTENANCE PRACTICES

Paragraph 8 is amended to read:

8. GMU 44/B Magnetometer

A. Remove the GMU 44/B Magnetometer

	Detail Steps/Work Items	Key Items/References
(1)	Make sure that the ELECT. MASTER switch is set to OFF.	
(2)	Remove the magnetometer assembly: <ul style="list-style-type: none"> – Remove the 3 screws that attach the magnetometer assembly to the lower surface of the right wing. – Lower the magnetometer assembly clear of the structure and disconnect the electrical cables. – Move the magnetometer assembly clear of the airplane. 	Refer to Figure 13 for GMU 44 and Figure 14 for GMU 44B. Support the assembly.
(3)	If necessary, remove the magnetometer from the panel.	

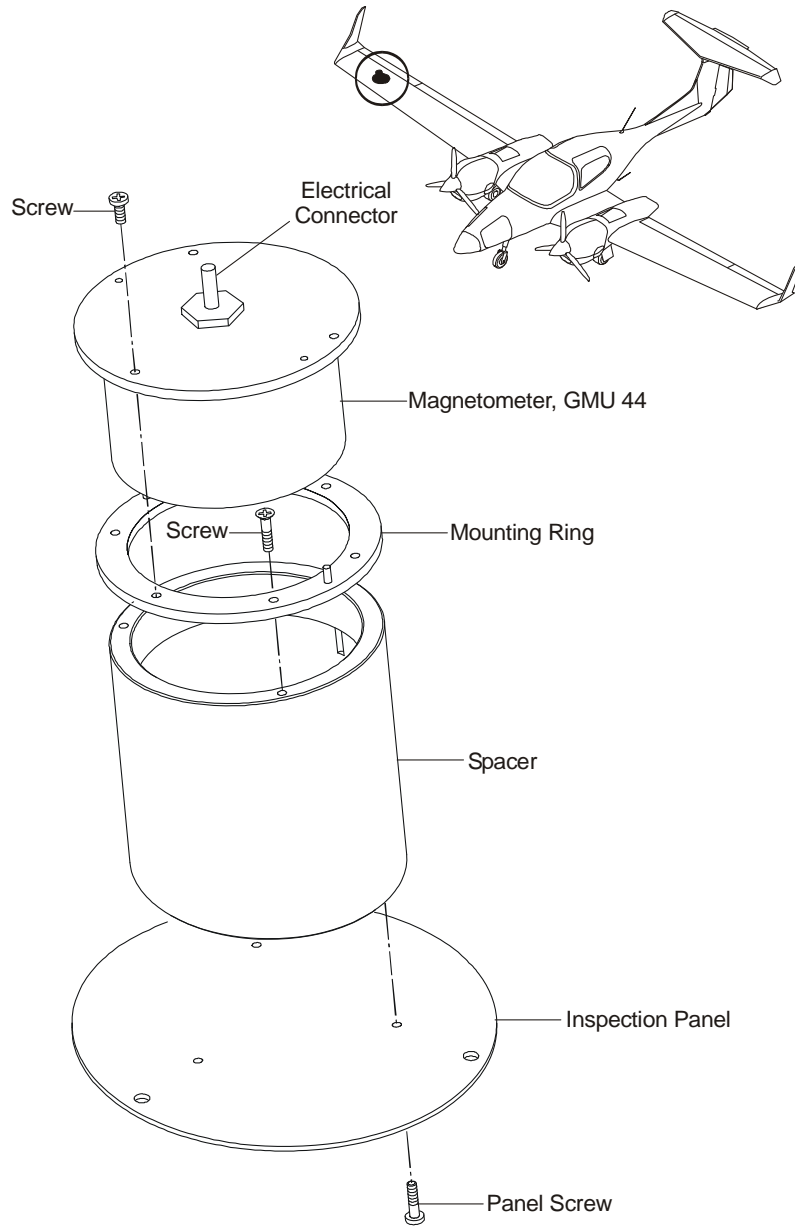


Figure 13: Magnetometer GMU 44 Assembly Installation

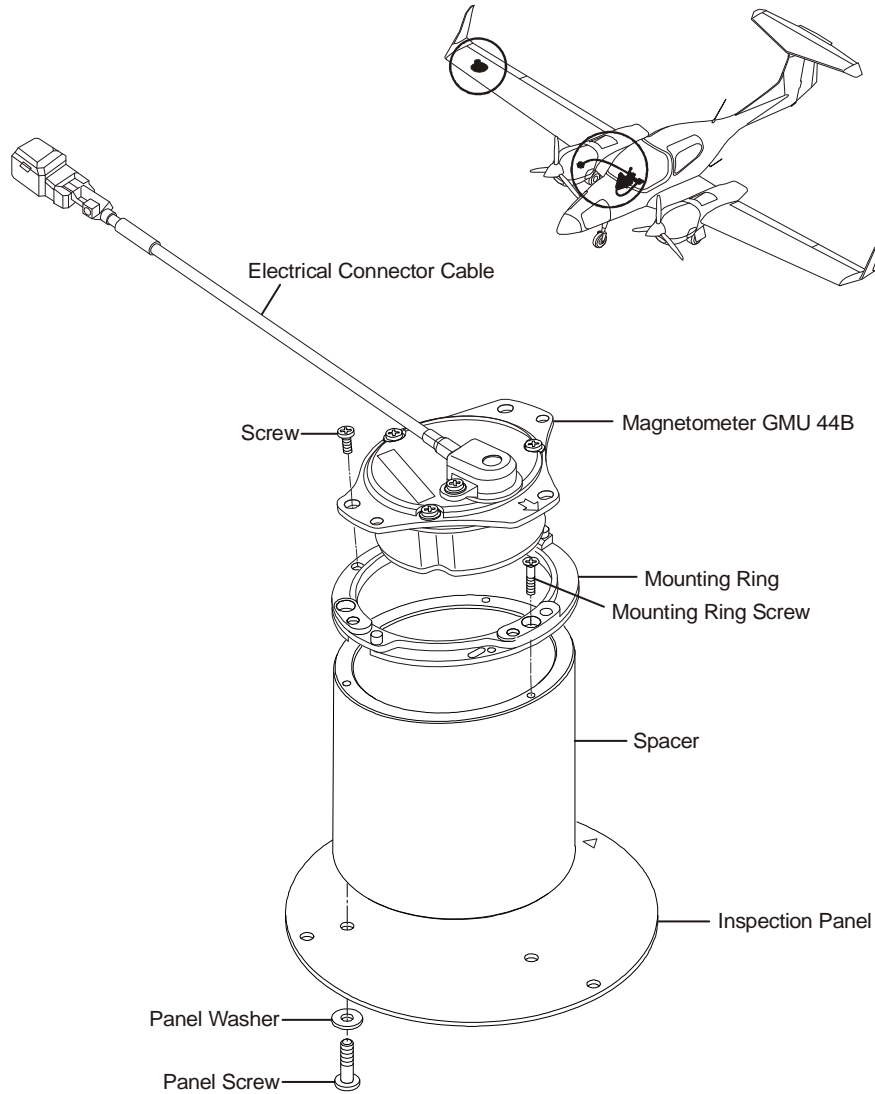


Figure 14: Magnetometer GMU 44B Assembly Installation

B. Install the GMU 44/B Magnetometer

	Detail Steps/Work Items	Key Items/References
(1)	Install the magnetometer assembly: <ul style="list-style-type: none">– Move the magnetometer assembly into position at the right wing.– Connect the electrical cables to the magnetometer assembly.– Move the magnetometer assembly fully into position in the lower surface of the right wing.– Install the 3 screws that attach the magnetometer assembly to the wing.	Pay attention on the dedicated mounting direction marked by an arrow! At the in-line connector.
(2)	Do a test for the correct operation of the integrated cockpit system (ICS): <ul style="list-style-type: none">– Set the ELECT. MASTER switch to ON.	The ICS must power up and successfully complete its selftest procedure.
(3)	Do a magnetometer calibration.	Refer to Paragraph 9.

CHAPTER 92
WIRING DIAGRAMS

1. General

The following item is amended to read:













Title	Drawing No.	Rev. No.	No. of Sheets
Schematic, G1000 NXi	D64-9231-60-04	E	6

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



REV	SH	ZONE	DESCRIPTION	DATE
"C"	ALL	ALL	MAM 42-1072 LRU designations updated.	23.03.2018
"D"	05	F1-3	MAM 42-1108 HSDB connection between PFD and GMA added.	04.12.2018
	05	B1-3	HSDB connection to GWX and GDL clarified.	
"E"	05	E/4	OAM 42-369 Optional Wiring for GMU 4B added.	08.10.2020


NOTES:

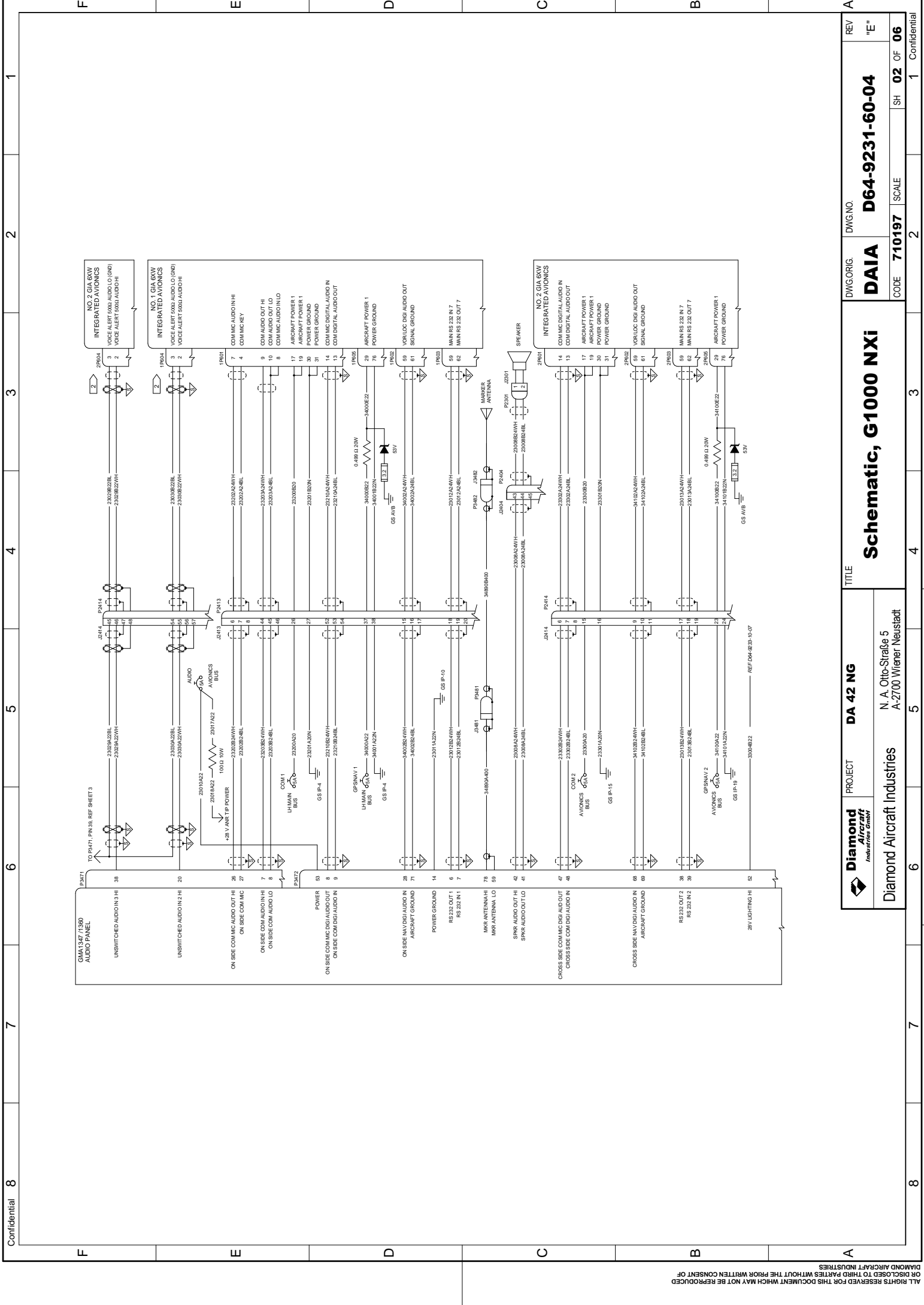
1. SYMBOL DESIGNATIONS

 TWISTED SHIELDED SINGLE CONDUCTOR SHIELD TERMINATED TO GROUND	 TWISTED SHIELDED SINGLE CONDUCTOR SHIELD FLOATS	 TWISTED SHIELDED PAIR SHIELD TERMINATED TO GROUND	 TWISTED SHIELDED PAIR SHIELD FLOATS	 TWISTED SHIELDED 3 CONDUCTOR SHIELD TERMINATED TO GROUND	 TWISTED SHIELDED 3 CONDUCTOR SHIELD FLOATS	 TWISTED SHIELDED 4 CONDUCTOR SHIELD TERMINATED TO GROUND	 TWISTED SHIELDED 4 CONDUCTOR SHIELD FLOATS	 AIRCRAFT GROUND	 GARMIN (SHIELD BLOCK) GROUND REFER TO DOC. 190-00313-09	 WIRE SPLICE CONNECTION	 COAXIAL CABLE	N/C = NO CONNECTION
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FLAG NOTES

1		OPTIONAL INSTALLATION IN REAR FUSELAGE, IF OAM 42-203 IS INSTALLED
2		USE DOUBLE-SHIELDED WIRING; BOTH SHIELDS SHOULD BE GROUNDED ON THE RECEIVING SIDE, ON THE TRANSMITTING SIDE, ONLY THE OUTER SHIELD SHOULD BE GROUNDED, WHILE THE INNER SHIELD SHOULD FLOAT.
3		WIRE / CABLE NOT INSTALLED WITH GTX 36R.
4		EITHER CABLE 34430B24 (TAS6XX) OR 34623A24 (GTX345R) ARE INSTALLED IN CONNECTOR P2414.

Diamond Aircraft Industries GmbH		Diamond Aircraft Industries		N. A. Otto-Straße 5 A-2700 Wiener Neustadt	
NAME / SIGN / DATE		PROJECT DA 42 NG		TITLE	
DRAWN:		CHECKED:		RELEASED:	
IDENTIFICATION MARKINGS DP-S-17-00001		CLASSIFICATION: NONE		INTERCHANGEABLE PART NO	
FIRST ANGLE PROJECTION 		FORMAT A3		FINISH IN MICRO-METER 3.2	
IF NOT OTHERWISE SPECIFIED GEOMETRIC DIMENSIONING AND TOLERANCING ACC. TO ISO 2768-mK		DIMENSIONS IN mm		SOFTWARE: SOLID EDGE ST6	
DIMENSIONS IN mm		FILENAME D64-9231-60-04d.dft		DWG. ORG. DAIA	
DIMENSIONS IN mm		DWG. NO. D64-9231-60-04		REV. "E"	
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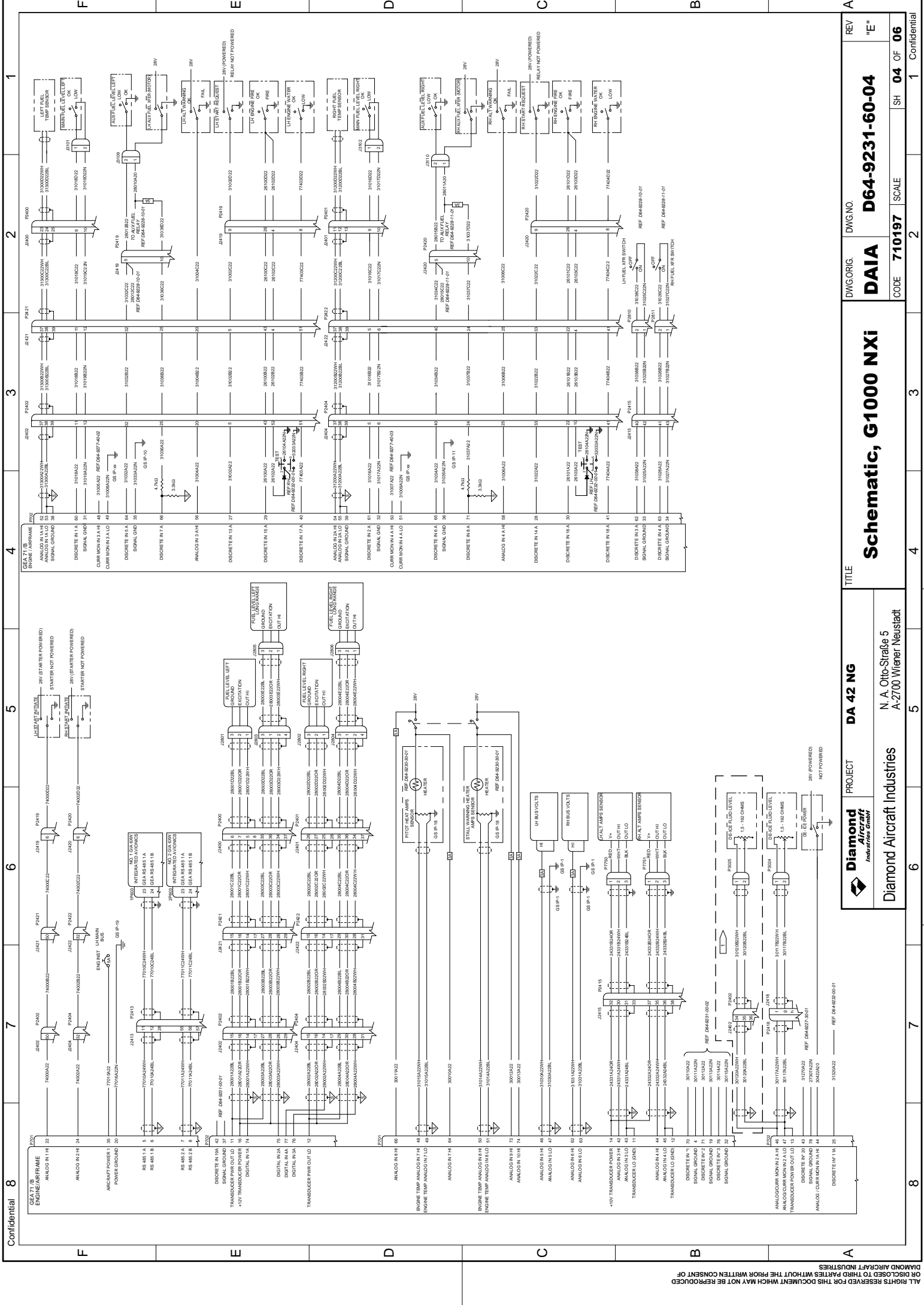


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B	DWG. ORIG.	DAIA
	CODE	710197
C	SCALE	1
	SH	02 OF 06
D	TITLE	Schematic, G1000 NXi
	PROJECT	DA 42 NG
E	Diamond Aircraft Industries GmbH	
	Diamond Aircraft Industries N. A. Otto-Straße 5 A-2700 Wiener Neustadt	
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PROJECT

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Diamond Aircraft Industries

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