

SERVICE BULLETIN



Service Bulletin No.: DAC1-30-02 Rev. 1

Date Issued: 25 August 2022

Title: Installation of a Heated Pitot Static Probe

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PRINT IN COLOUR

- 1. ATA Code:** 3030
- 2. Effectivity:** All DA20-C1 aircraft not equipped with recognition lights.
- 3. General:** This Service Bulletin addresses the installation of the Electric Pitot Heating System. A Heated Pitot Probe Assembly replaces the existing Pitot probe and the associated wiring is installed across the instrument panel, fuselage and wing.
- 4. Compliance:** Compliance with this Service Bulletin is optional.
- 5. Approval:** Engineering data referenced or contained in this Service Bulletin is approved as part of the type design.
- 6. Labour:** Approximately 20 hours will be required to accomplish this Service Bulletin.
This estimate is for direct labor performed by a technician and it does not include setup, planning, familiarization, cure time, part fabrication or tool acquisition.
- 7. Material:** The following list of materials may be ordered as kit: **DAC1-30-02-AMK0.**

Part Number	Description	Qty
31886	Ring Terminal, Red, 16-22 AWG, Stud Size 8	1
35108	Ring Terminal	2
36151	Ring Terminal, Red, 16-22 AWG, Stud Size 6	1
36157	Ring Terminal, Blue, 14-16 AWG, Stud Size 6	2
1-480318-0	MATE-N-LOK Receptacle, 2 Positions	1
1N4007	Diode, Rectifier	1
* 20-3030-02-01	Pitot Wire Conduit	1
206062-3	Cable Clamp Kit, Shell Size 11, CPC	2
206403-1	Peripheral Seal, Shell Size 11, CPC	1
206429-1	CPC Connector, Plug, 4 Positions	1
206430-2	CPC Connector, Receptacle, 4 Positions	1
22-3030-00-09	Gasket, Outer, Phenolic	1
22-3030-00-11	Gasket, Inner, Phenolic	1
22-3030-01-00	Heated Pitot Static Probe Sub Assembly	1
22-3030-02-00	Current Monitor	1
22-3030-60-00	Bracket Assembly, Limit Switch	1
2-520184-2	Faston, Red, 18-22 AWG	5
330369	Splice, Closed Barrel, 10-12 AWG	5
3-350820-2	Faston, Blue, 14-16 AWG	1
3PS-125-2W	Wire Marking Sleeve, 2", White, 16-22 AWG	A/R
3PS-250-2WT-2	Wire Marking Sleeve, 2", White, 10-16 AWG	A/R
3PS-500-2W	Wire Marking Sleeve, 2", White, 1-8 AWG	2

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4391S14-209	Limit Switch, Manual Reset	1
60617-1	MATE-N-LOK Socket, Tin Plated, 18-24 AWG	2
640907-1	Faston, Yellow, 10-12 AWG	2
66103-4	Pin, Yellow, Gold Plated, 20-24 AWG	3
66105-4	Socket, Yellow, Gold Plated, 20-24 AWG	3
66360-4	Socket, Violet, Gold Plated, 14-18 AWG	6
66361-4	Pin, Violet, Gold Plated, 14-18 AWG	6
920-1-4 BKWH	Expando Sleeving, 1/4"	1.9m
AN3-4A	Bolt	1
AN526-632R24	Screw, Pan Head	3
ATUM-1-4-0	Heat Shrink Tubing, 1/4"	A/R
ATUM-1-8-0	Heat Shrink Tubing, 1/8"	A/R
CCR244SS3-04	Cherry Rivet	1
DP410	Epoxy Adhesive, 3M	A/R
EMS-A-C0	Cable Tie Base	2
HTCP-20S	Heat Transfer Compound	A/R
K1ACAPCADA	Rocker Switch, White, with Indicator	1
M22759/16-10-9	Wire, 10 AWG	100mm
M22759/16-14-9	Wire, 14 AWG	A/R
M22759/16-20-9	Wire, 20 AWG	A/R
M22759/16-22-9	Wire, 22 AWG	A/R
M22759/34-10-9	Wire, 10 AWG	A/R
M24236/1-0280	Limit Switch	1
MS21042-06	Nut, Self Locking	3
MS21044N3	Nut, Self Locking	1
MS3320-15	Circuit Breaker, 15 AMP	1
MS3367-1-9	Cable Tie (203mm x 4.7mm)	A/R
MS3367-5-9	Cable Tie (142mm x 3.6mm)	A/R
MS51957-44	Pan Head Screw	2
NAS1149C0363R	Washer, Flat	2
NAS1149DN816H	Washer, Thin	2
NAS1149FN216P	Washer, Flat	1
NAS1149FN632P	Washer, Flat	3
RAYRIM NR.6	Adhesive Lined, Protective Edging	40mm
RNF100-1-4 WHT	Heat Shrink Tubing, 1/4", White	A/R
RNF100-1-8 CLEAR	Heat Shrink Tubing, 1/8", Clear	A/R
VF4-65F11-S01	Relay	1

* Pitot Wire Conduit is not required with MOD 0343 or OÄM 22-0122 installed.

Kit number DAC1-30-03-AMK1 can be supplied for customers who do not need that conduit.

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8. Special Tools: N/A

9. References: DA20-C1 Aircraft Maintenance Manual (AMM), Document Number DA201-C1.

10. Accomplishment Instructions

10.1 The subsequent instructions have been broken up into three aircraft sections; Instrument Panel, Fuselage and Wing. Each section contains Modification Instructions and Wiring Instructions. The Modification Instructions detail the physical changes to the existing aircraft whereas the Wiring Instructions detail the installation of electrical components and routing of new wires. At times these areas overlap. Instructions for closing the aircraft have been included at the end.

11. Accomplishment Instructions: INSTRUMENT PANEL

11.1 Instrument Panel Modification

- 11.1.1** Disconnect the Aircraft Battery according to AMM # DA201-C1, Chapter 24-31.
- 11.1.2** Remove the Instrument Panel Cover according to AMM # DA201-C1, Chapter 25-10.
- 11.1.3** On the right side of the instrument panel, open the BATTERY Circuit Breaker on the ELECTRICAL Circuit Breaker panel.
- 11.1.4** Assembly Circuit breaker Jumper as shown in Figure 1.



Figure 1: Circuit Breaker Jumper

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11.1.5 Add the PITOT HEAT Circuit Breaker:

11.1.5.1 Non-G500: On the SYSTEM Circuit Breaker Panel, remove the circuit breaker cap at the PITOT HEAT location and install the 15A Circuit Breaker (p/n: MS3320-15), as shown in Figure 2.

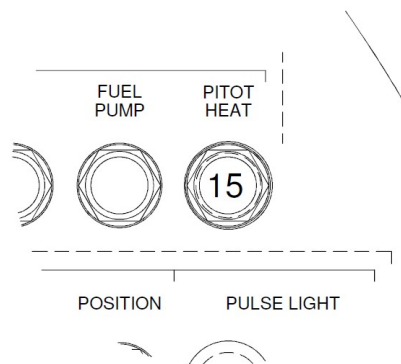


Figure 2: Circuit Breaker for Non-G500 Instrument Panel

11.1.5.2 G500: Remove the circuit breaker cap directly below the LANDING circuit breaker and install the 15A Circuit Breaker (p/n: MS3320-15). Add the label "SYSTEM" and "PITOT HEAT" to the placard, as shown in Figure 3.

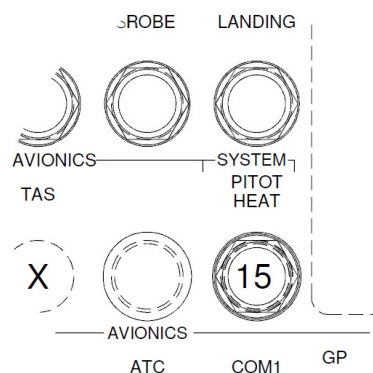


Figure 3: Circuit Breaker for G500 Instrument Panel

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- 11.1.6** Connect Circuit Breaker Jumper to top side of PITOT HEAT circuit breaker and the other side to the SYSTEM BUS.
- 11.1.7** Remove the section of placard located over the switch opening and replace the existing "PULSE" text with "PITOT HEAT", as shown in Figure 4.

Replace "PULSE" with "PITOT HEAT"

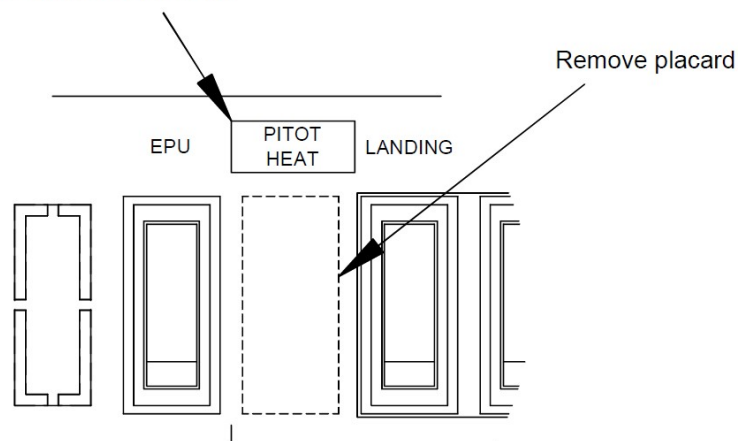


Figure 4: LIGHTS Panel

- 11.1.8** Remove the panel insert and file smooth as required, as shown in Figure 5. Install the Rocker Switch (p/n: K1ACAPCADA).

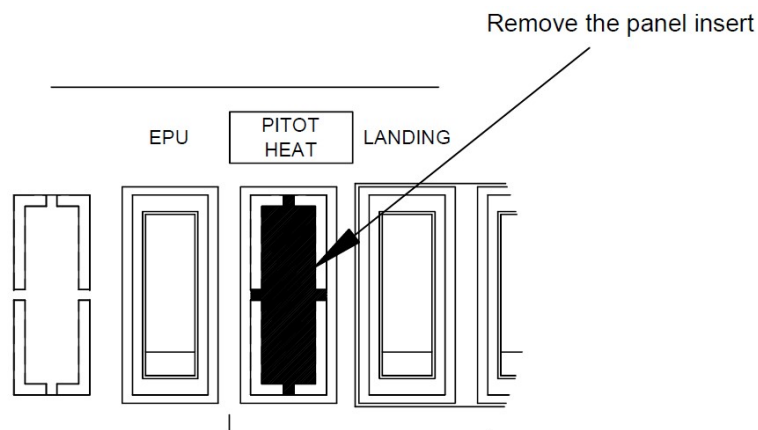


Figure 5: PITOT HEAT Rocker Switch

11.2 Instrument Panel Wiring Installation

11.2.1 Follow the instructions below to install the Current Monitor (p/n: 22-3030-02-00), as shown in Figure 6.

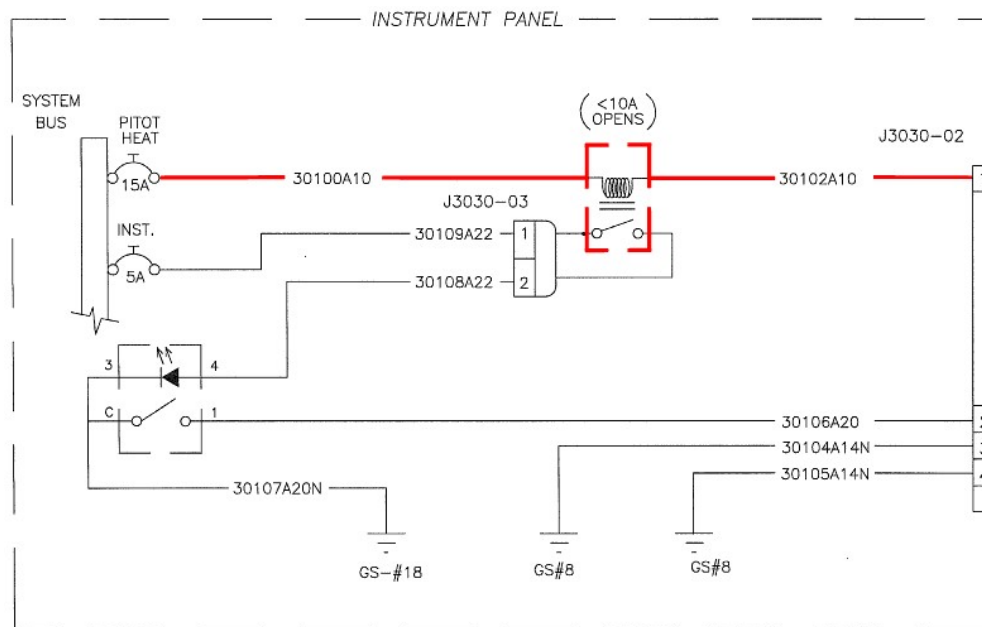


Figure 6: Instrument Panel Schematic highlighting the Current Monitor Installation

11.2.1.1 Modify the wires for the Current Monitor (p/n: 22-3030-02-00) according to Figure 7.

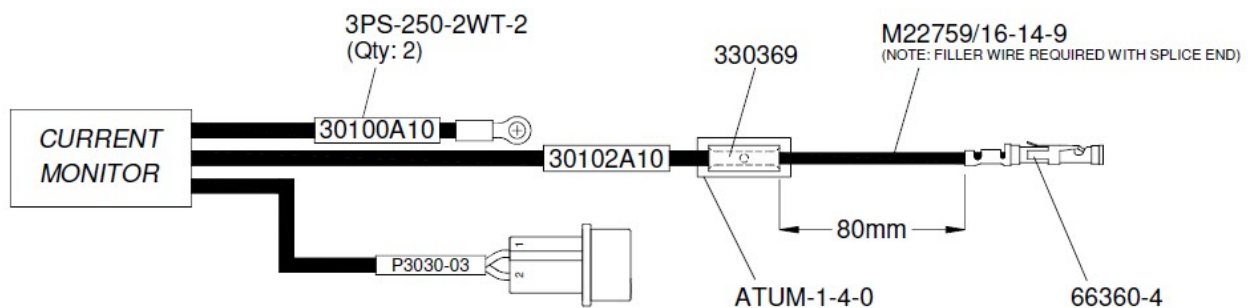


Figure 7: Modification of Wires for Current Monitor

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11.2.1.2 Secure the current monitor to the instrument panel using the supplied cable ties (p/n: MS3367-5-9). Route the wires as shown in Figure 8 or Figure 9.

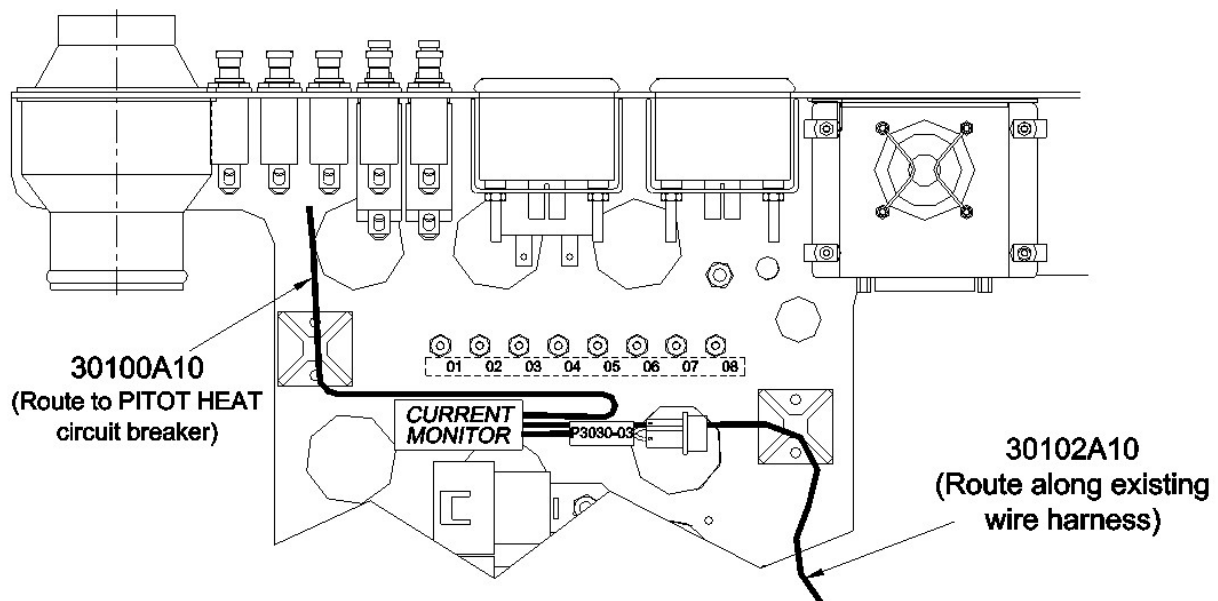


Figure 8: Current Monitor for Non-G500 Instrument Panel

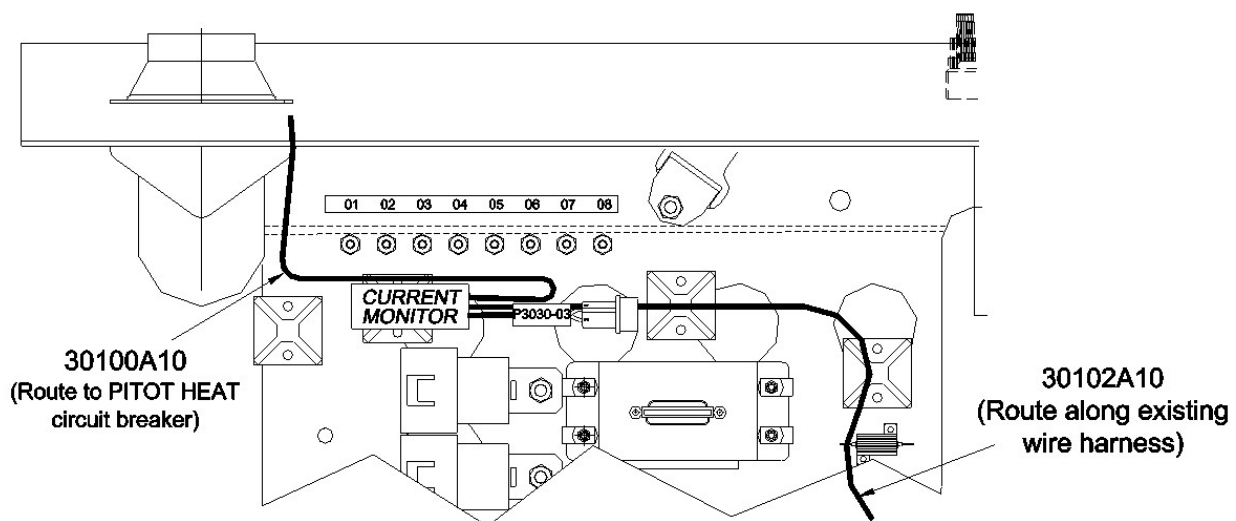


Figure 9: Current Monitor for G500 Instrument Panel

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11.2.1.3 Connect wire 30100A10 to the PITOT HEAT circuit breaker.

11.2.2 Follow the instructions below to install the J3030-03 connection with the Current Monitor, as shown in Figure 10.

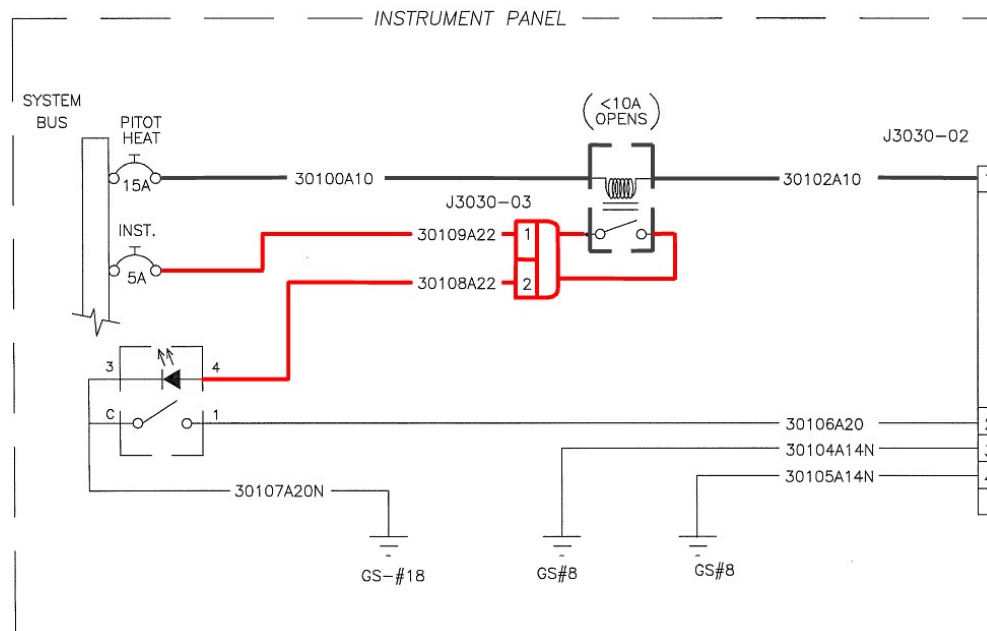


Figure 10: Instrument Panel Schematic highlighting the Current Monitor Connection

11.2.2.1 Prepare the wires for the J3030-03 Receptacle (p/n: 1-480318-0) according to Figure 11.

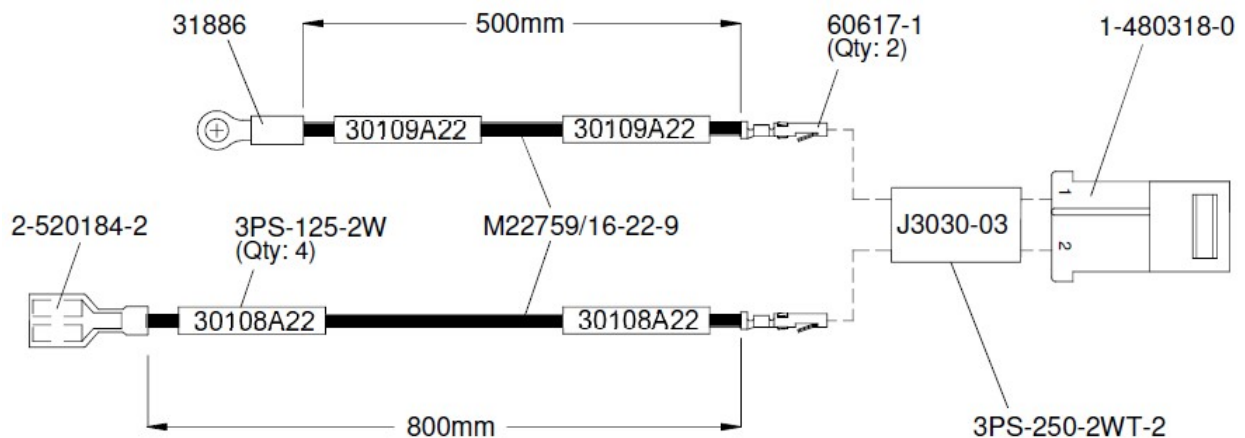


Figure 11: Preparation of Wires for J3030-03 Receptacle

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11.2.2.2 Plug together the J3030-03 connector and the P3030-02 connector.
Route the prepared wires as shown in Figure 12 or Figure 13.

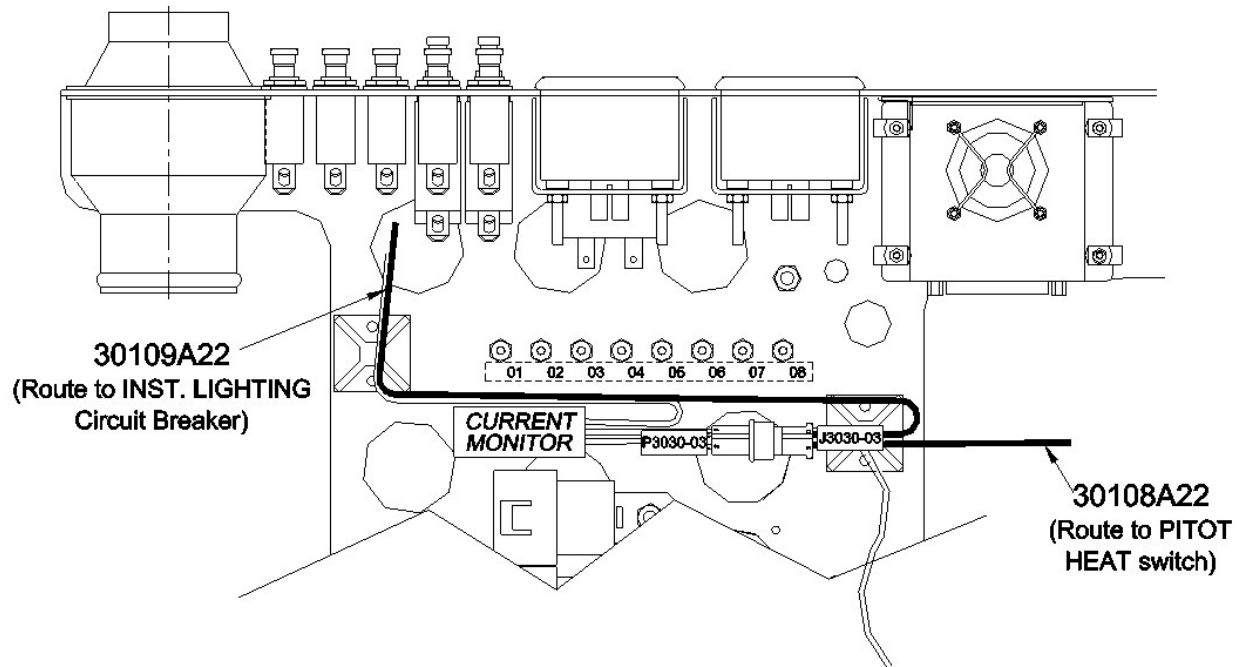


Figure 12: Routing of J3030-03 Connector for Non-G500 Instrument Panel

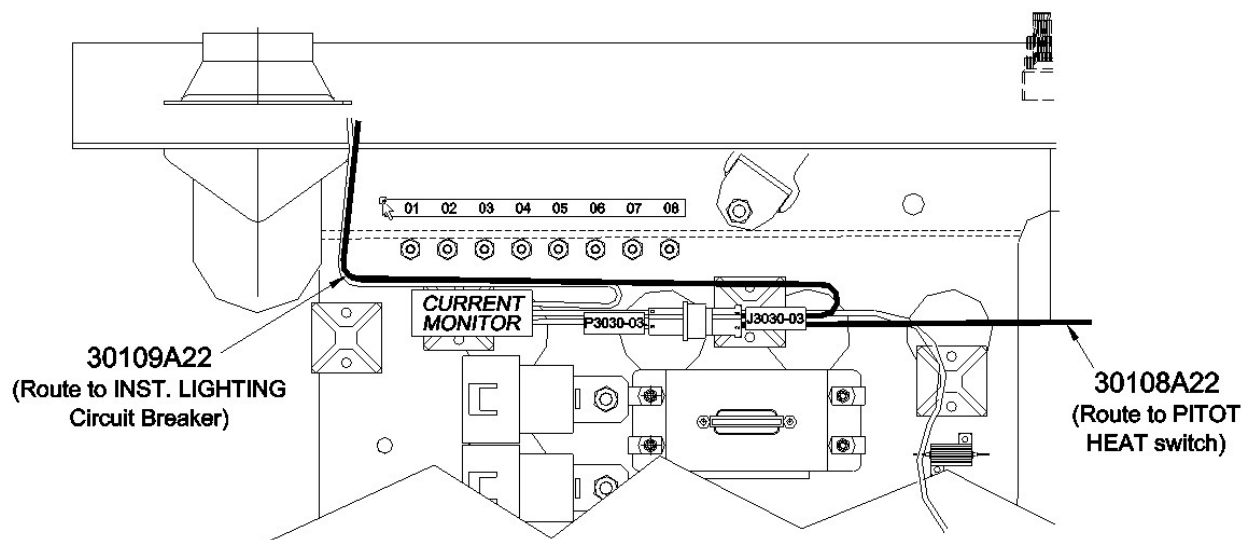


Figure 13: Routing of J3030-03 Connector for G500 Instrument Panel

11.2.2.3 Connect wire 30109A22 to the existing INSTRUMENT LIGHTING Circuit Breaker.

11.2.3 Follow the instructions below to install the PITOT HEAT Rocker Switch connections, as shown in Figure 14.

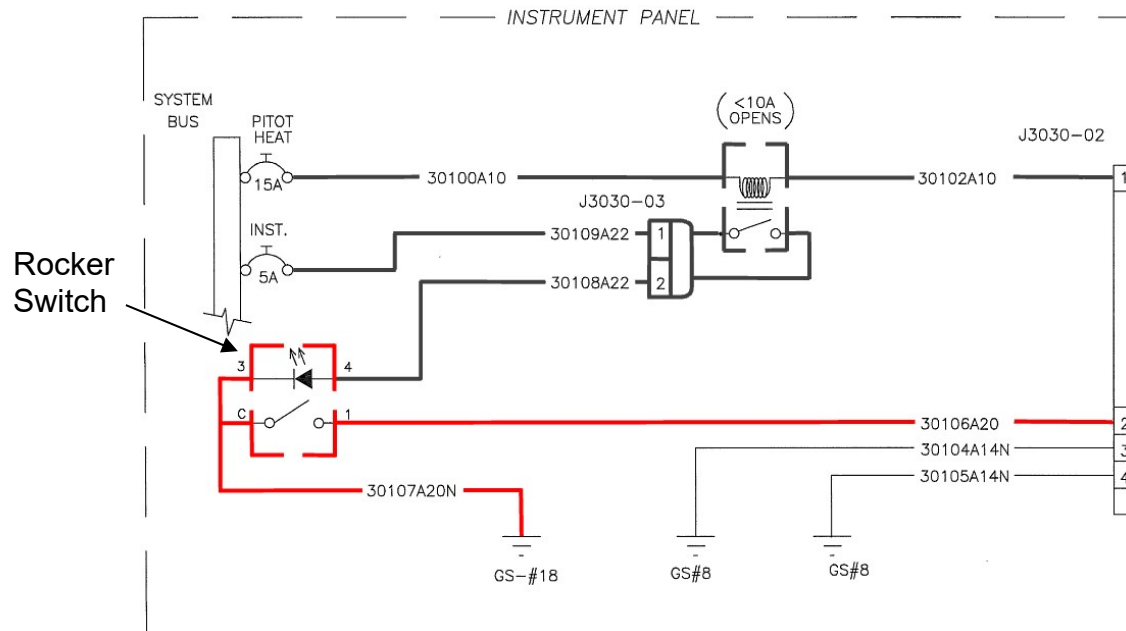


Figure 14: Instrument Panel Schematic highlighting the Rocker Switch Installation

11.2.3.1 Prepare the following lead wire according to Figure 15.

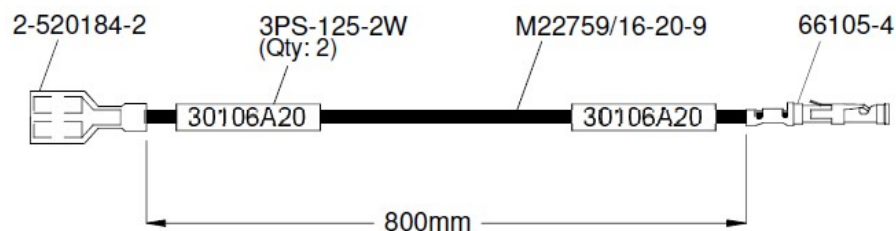


Figure 15: Preparation of Lead Wire for the PITOT HEAT Rocker Switch

11.2.3.2 Prepare the ground wire according to Figure 16.

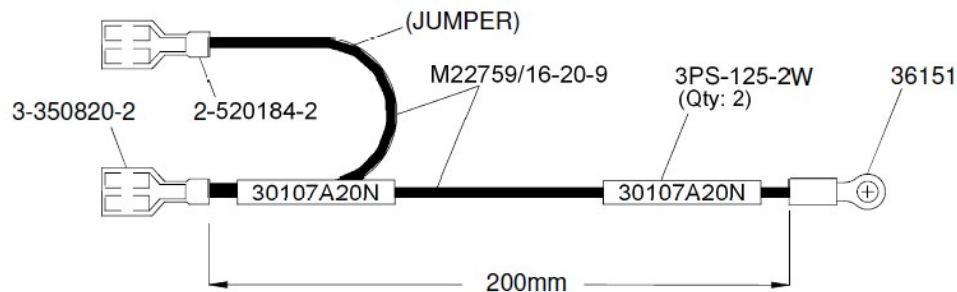


Figure 16: Preparation of Ground Wire for the PITOT HEAT Rocker Switch

11.2.3.3 Connect wires 30108A22, 30106A20, 30107A20N/JUMPER and JUMPER to the PITOT HEAT Rocker Switch, as shown in Figure 17.

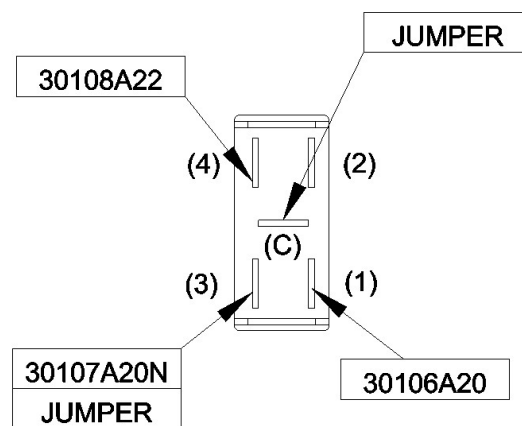


Figure 17: PITOT HEAT Rocker Switch Connections

11.2.3.4 Route and secure wire 30107A20N to ground stud #18, located forward of the PITOT HEAT Rocker Switch.

11.2.3.5 Route wire 30106A20 along the existing instrument panel wire harness to 30102A10 (See Figure 8 or 9).

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- 11.2.4 Follow the instructions below to assemble the wire connection for the instrument panel to the fuselage, as shown in Figure 18.

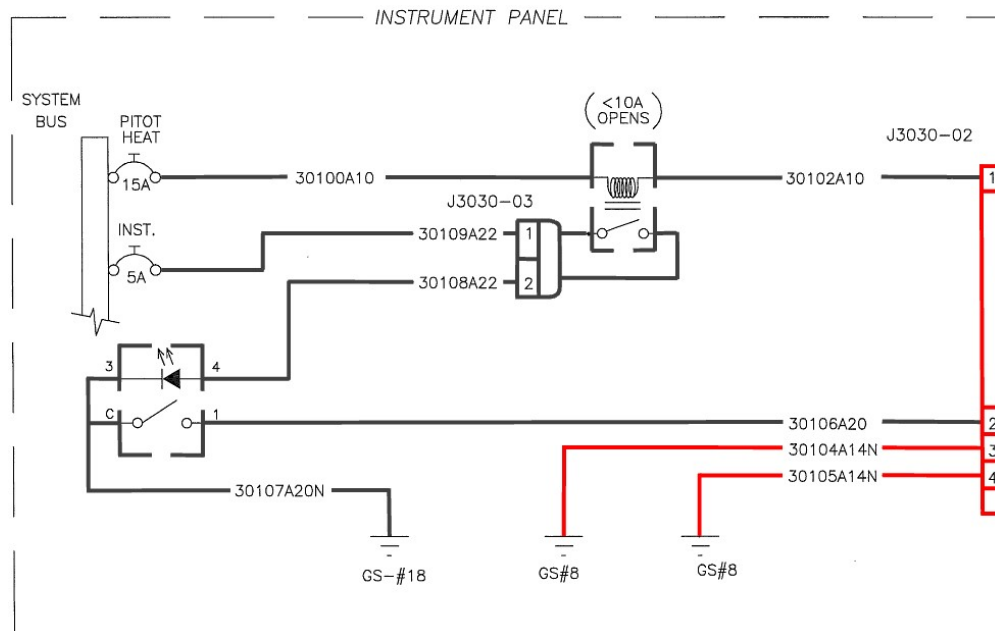


Figure 18: Instrument Panel Schematic highlighting the Fuselage Connection

- 11.2.4.1 Prepare the ground wires according to Figure 19.

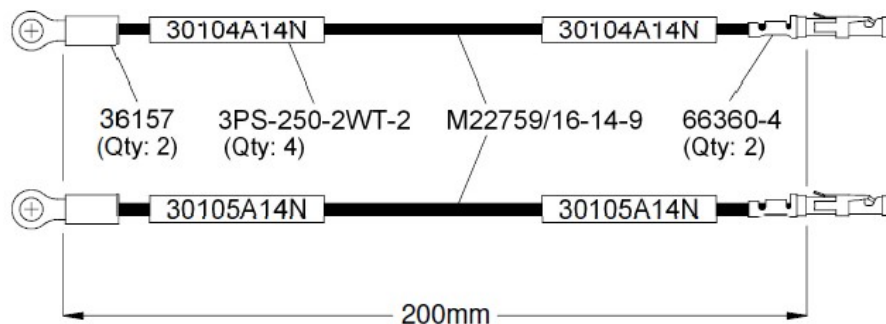


Figure 19: Preparation of Ground Wires for the Instrument Panel

- 11.2.4.2 Route and secure the ground wires 30104A14N and 30105A14N to ground stud #8.

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11.2.4.3 Install wires 30102A10, 30106A20, 30104A14N and 30105A14N into the J3030-02 Receptacle, according to Figure 20.

11.2.4.4 Install the Peripheral Seal (p/n: 206403-1) into the J3030-02 receptacle using Loctite 495 or equivalent, according to Figure 20.

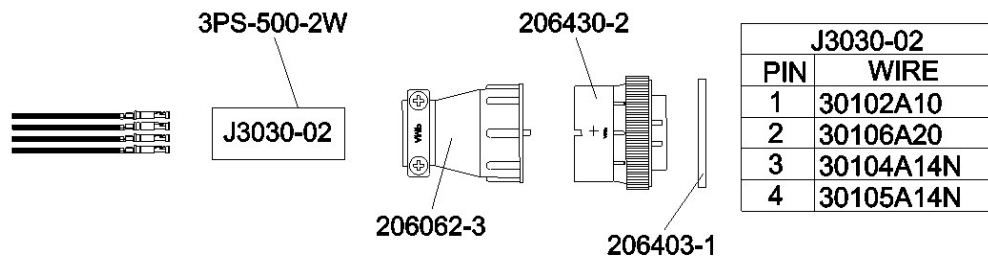


Figure 20: Instrument Panel J3030-02 Receptacle

12. Accomplishment Instructions: FUSELAGE

12.1 Fuselage Modification

12.1.1 Remove the Pilot Seat according to AMM # DA201-C1, Chapter 25-10.

12.1.2 Drill a 5mm diameter hole in the forward control Bulkhead as shown in VIEW A of Figure 21.

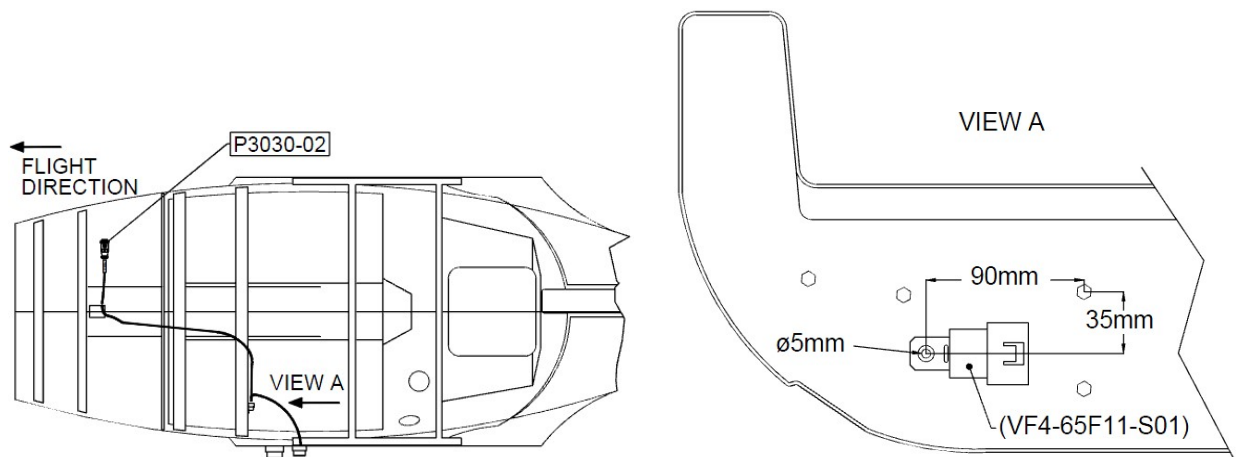


Figure 21: Modification to Forward Control Bulkhead

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12.2 Fuselage Wiring Installation

- 12.2.1 Follow the instructions below to assemble the wire connection for the fuselage to the instrument panel, as shown in Figure 22.

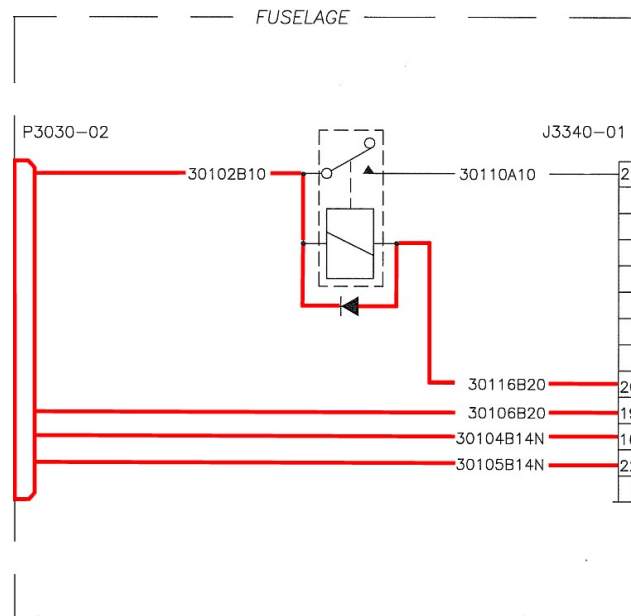


Figure 22: Fuselage Schematic highlighting the Instrument Panel Connection

- 12.2.1.1 Prepare the relay wires according to Figure 23.

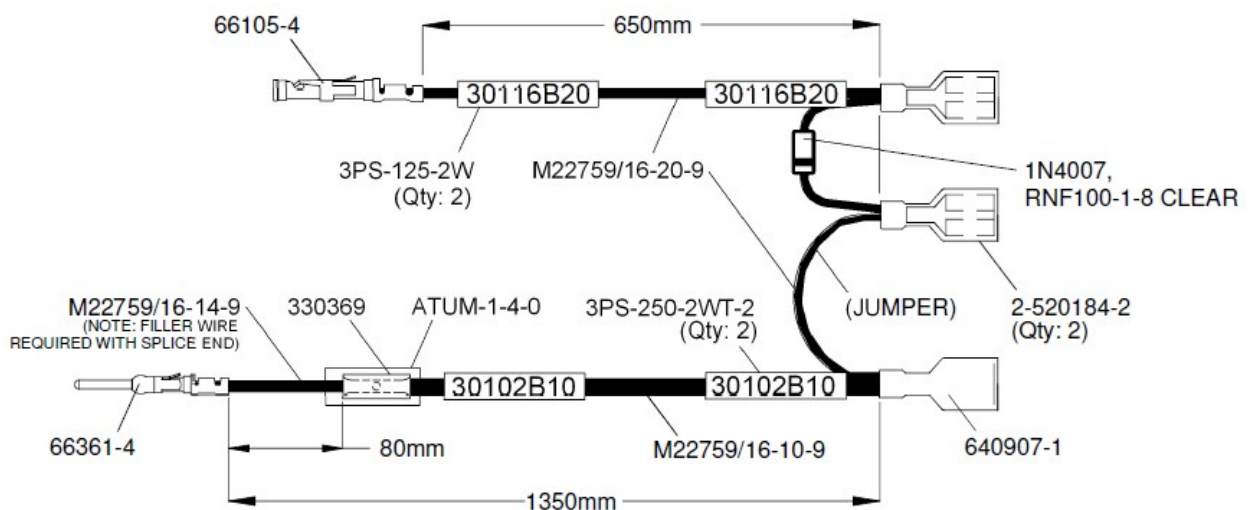


Figure 23: Preparation of Fuselage Relay Wires

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12.2.1.2 Prepare the lead wire according to Figure 24.

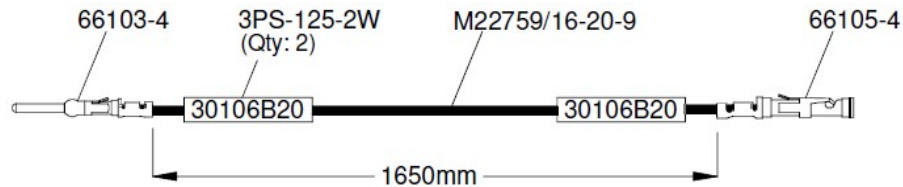


Figure 24: Preparation of Fuselage Lead Wire

12.2.1.3 Prepare the ground wires according to Figure 25.

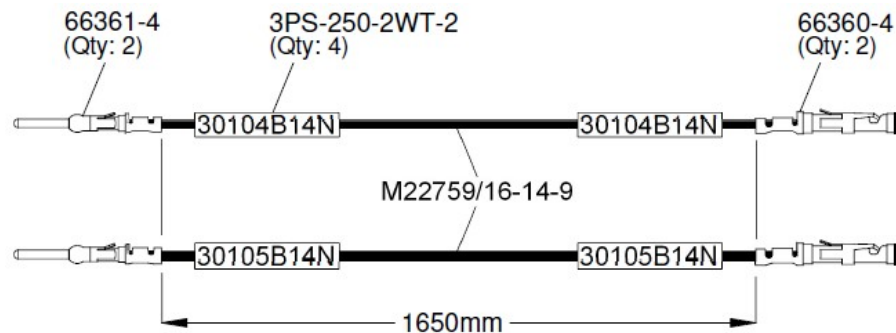


Figure 25: Preparation of Fuselage Ground Wires

12.2.1.4 Install wires 30102B10, 30106B20, 30104B14N and 30105B14N into the P3030-02 Plug, according to Figure 26.

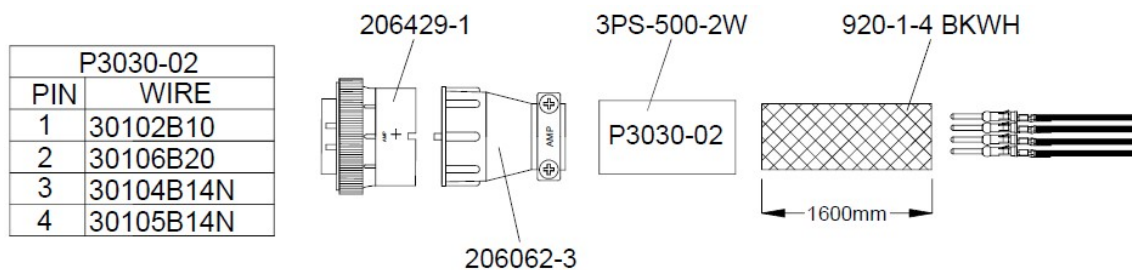


Figure 26: Fuselage P3030-02 Plug

12.2.2 Connect the fuselage P3030-02 Plug with the instrument panel J3030-02 Receptacle.

12.2.3 Route the new wire harness along the existing fuselage wire harness and secure with the supplied cable ties (p/n: MS3367-5-9). See Figure 21 as reference.

NOTE: Throttle quad may need to be removed in order to properly secure the wire harness.

12.2.4 Follow the instructions below to install the relay and associated connections, as shown in Figure 27.

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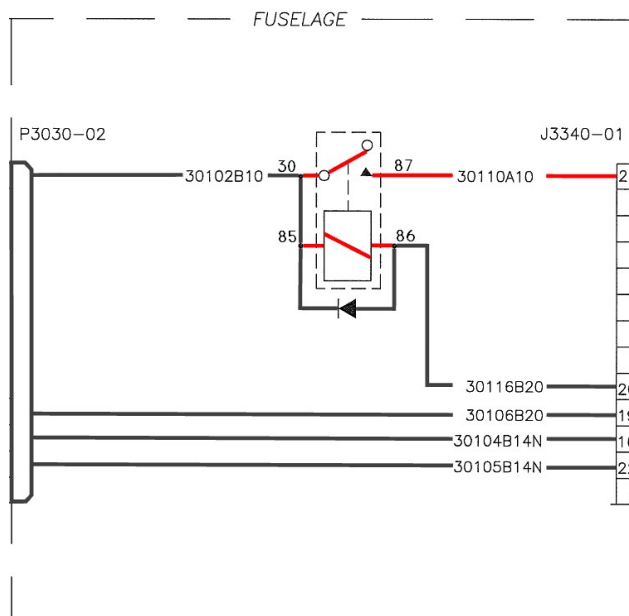


Figure 27: Fuselage Schematic highlighting the Relay Installation

12.2.4.1 Mount the relay in the Forward Control Bulkhead according to Figure 28. Reference Figure 21 for the location of the hole and the orientation of the relay.

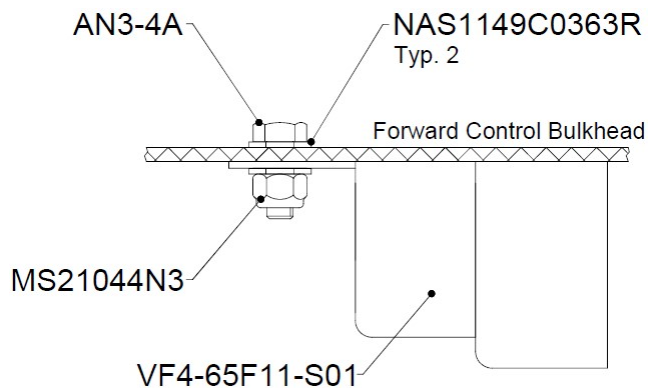


Figure 28: Mounting the Relay

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12.2.4.2 Prepare the remaining relay wire according to Figure 29.

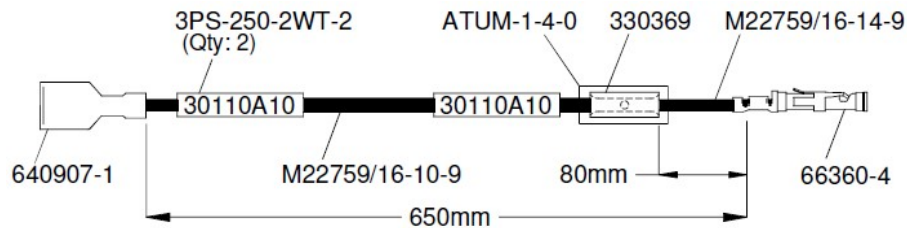


Figure 29: Wire Preparation for the Relay

12.2.4.3 Connect wires 30110A10, 30102B10, 30116B20/JUMPER and JUMPER to the Relay according to Figure 30.

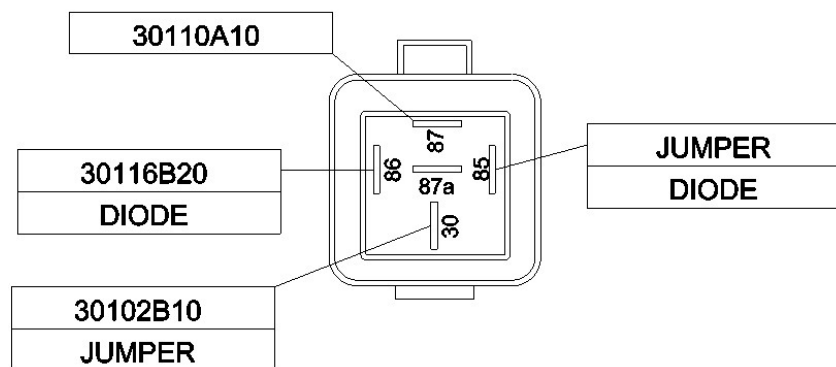


Figure 30: Relay Connections

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- 12.2.5** Follow the instructions below to assemble the wire connection for the fuselage to the wing, as shown in Figure 31.

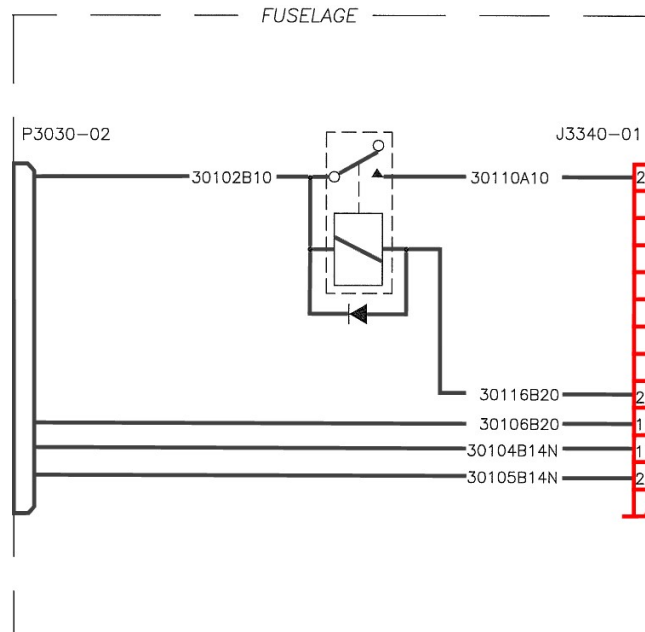


Figure 31: Fuselage Schematic highlighting the Wing Connection

- 12.2.5.1 Route wires 30110A10, 30116B20, 30106B20, 30104B14N and 30105B14N to the existing fuselage J3340-01 Receptacle located at the left-hand wing root under the pilot seat.
- 12.2.5.2 Install the wires into the existing fuselage J3340-01 Receptacle according to Figure 32.

J3340-01	
PIN	WIRE
21	30110A10
20	30116B20
19	30106B20
16	30104B14N
22	30105B14N

Figure 32: Fuselage Receptacle for Wing Plug

- 12.2.5.3 Secure the wires to the existing fuselage wire harness using the supplied cable ties (p/n: MS3367-5-9).

13. Accomplishment Instructions: WING

13.1 Wing Modification

NOTE: For aircraft that have the Heated Pitot probe provisions installed (MOD 0343 or OÄM 22-0122), only steps 13.1.3 and 13.1.4 need to be performed

13.1.1 Remove the LH Wing according to AMM # DA201-C1, Chapter 57-10.

13.1.2 Gain access to the left wing (aft of spar section) through the outboard Access Hole, as shown in Figure 33.

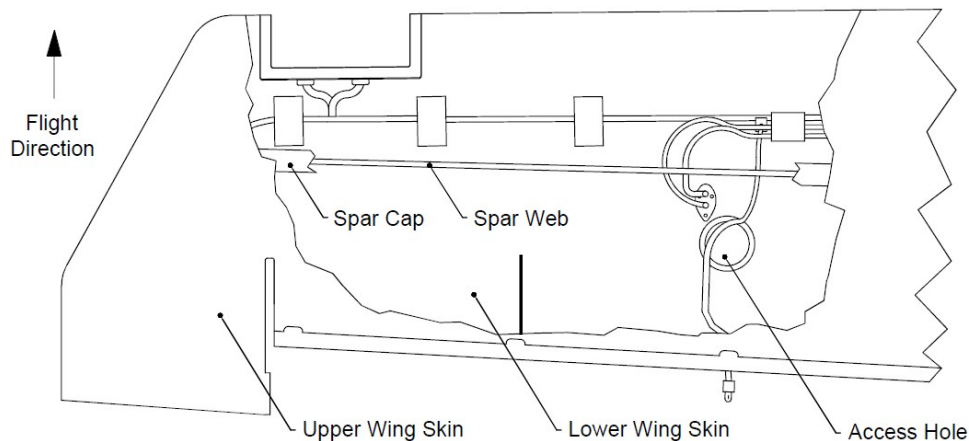


Figure 33: Left Wing Access Hole

13.1.3 Remove the Pitot Probe according to AMM # DA201-C1, Chapter 34-10.

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- 13.1.4** Modify the existing Pitot Probe Opening by removing the shaded area according to Figure 34. See Appendix A for a scaled template.

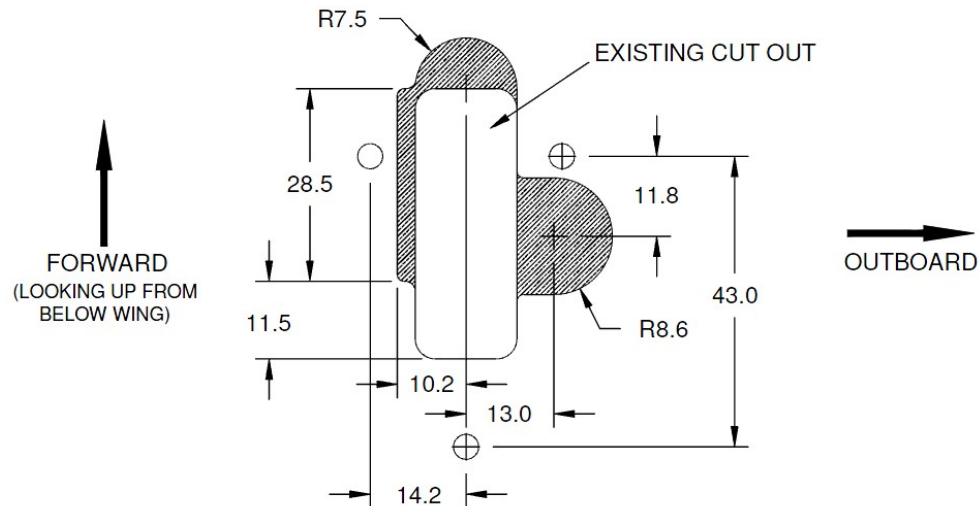


Figure 34: Modification to Pitot Probe Opening in Wing

- 13.1.5** Remove the Landing Light and Taxi Light according to AMM # DA201-C1, Chapter 33-40.
- 13.1.6** Gain access to the outboard left wing (forward of spar section) through the Landing/Taxi Light Cut-Out, as shown in Figure 35.

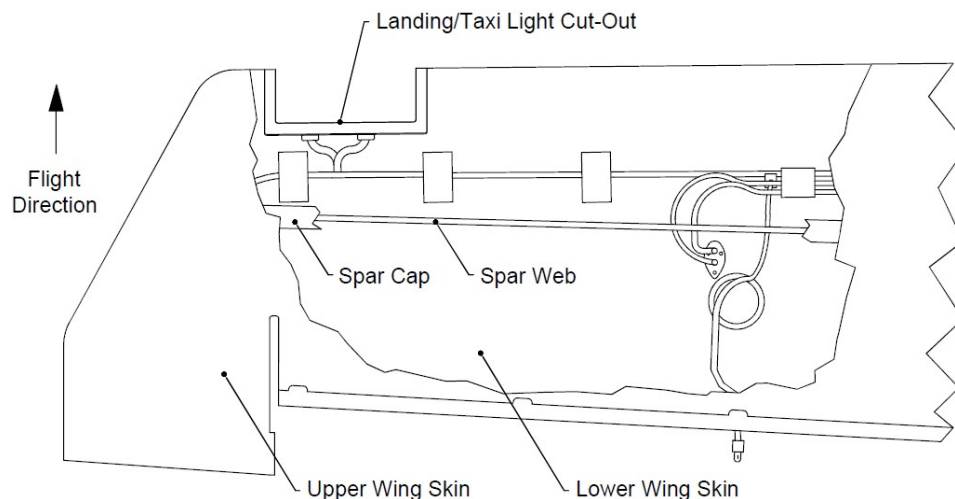


Figure 35: Left Wing Access through Landing/Taxi Light Cut-Out

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- 13.1.7** Enlarge the hole in the Wing Conduit to $\varnothing 13\text{mm}$ and shrink the Adhesive Lined, Protective Edging (p/n: RAYRIM NR.6) around the hole by applying heat, as shown in Detail G of Figure 36. Take caution as to not damage the existing wire harness for the wing.

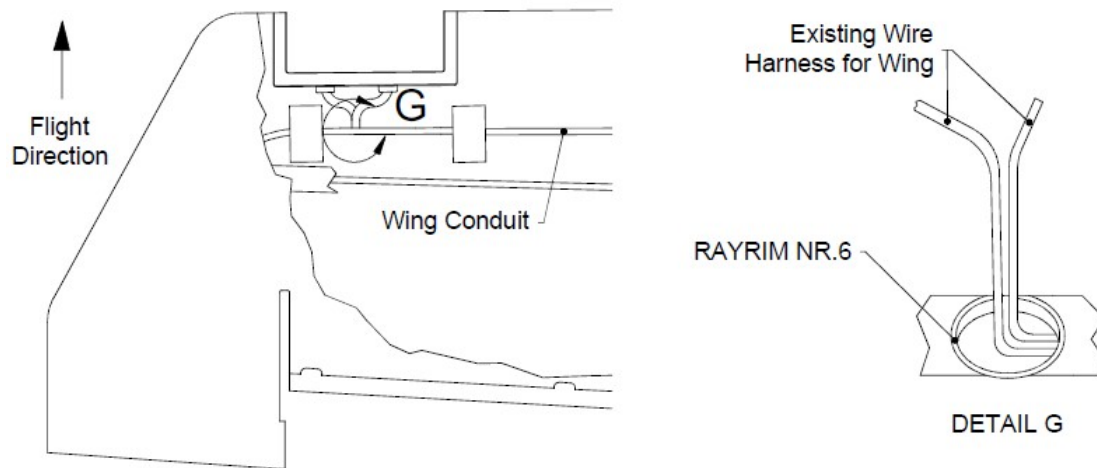


Figure 36: Modification to Wing Conduit

- 13.1.8** Drill a hole through the Spar Web according to View H of Figure 37. Take caution as to not damage the surrounding laminate.

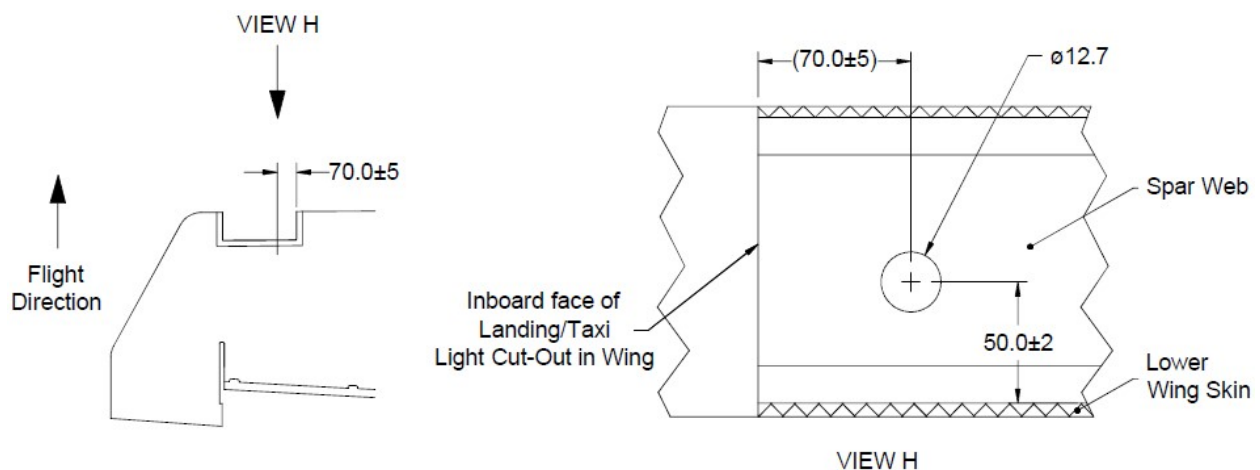


Figure 37: Modification to Wing Spar

- 13.1.9** Clean the work area and prepare the newly drilled hole in the Spar Web for bonding.

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- 13.1.10** Place the Pitot Wire Conduit (p/n: 20-3030-02-01) in the wing such that the straight end of the conduit rests on the lower wing skin near the Access Hole and the curved end of the conduit extends through the newly drilled hole, as shown in Figure 38.

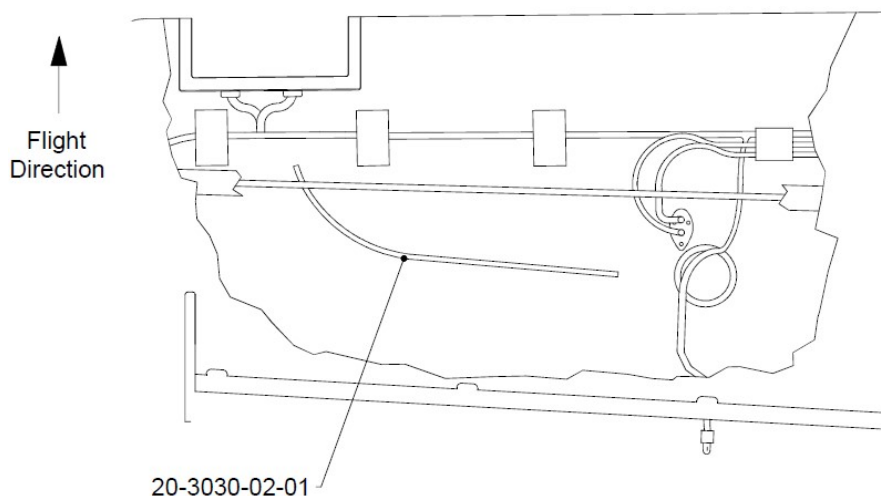


Figure 38: Location of Pitot Wire Conduit in Wing

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- 13.1.11** Bond the Pitot Wire Conduit through the newly drilled hole in the Spar Web using Epoxy Adhesive DP410, according to Detail J of Figure 39. Ensure the bonded area is free of voids and the straight end of the Pitot Wire Conduit rests on the Lower Wing Skin.

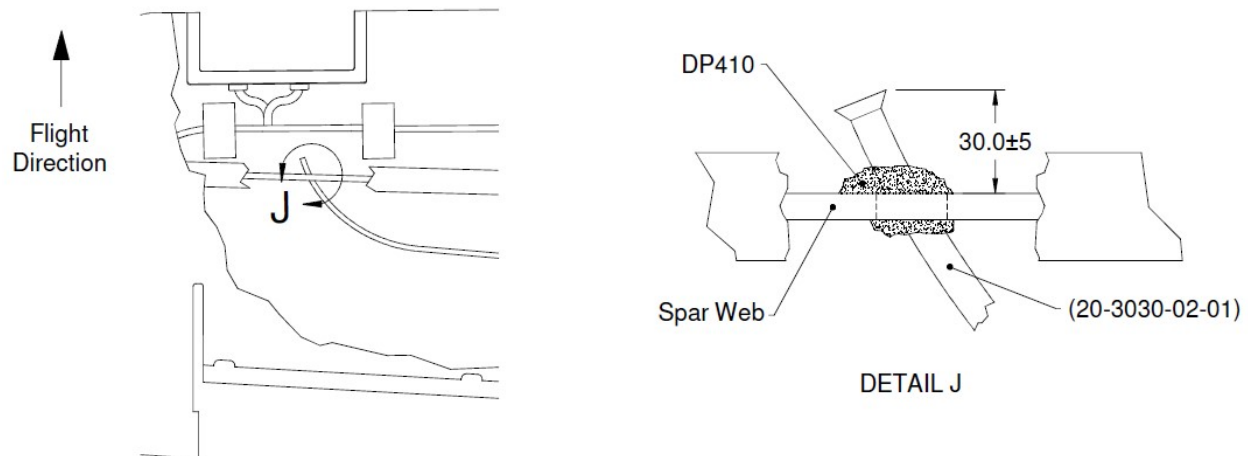


Figure 39: Bonding of Pitot Wire Conduit to Wing Spar

- 13.1.12** Bond the Cable Tie Base (p/n: EMS-A-C0) to the lower wing skin using Epoxy Adhesive DP410, as shown in Section K-K of Figure 40 (allow to cure). Fasten using Cable Tie (p/n: MS3367-1-9). Fill gaps around the Cable Tie and the Pitot Wire Conduit (20-3030-02-01) with DP410. Ensure the bonded area is free of voids and wipe away excess.

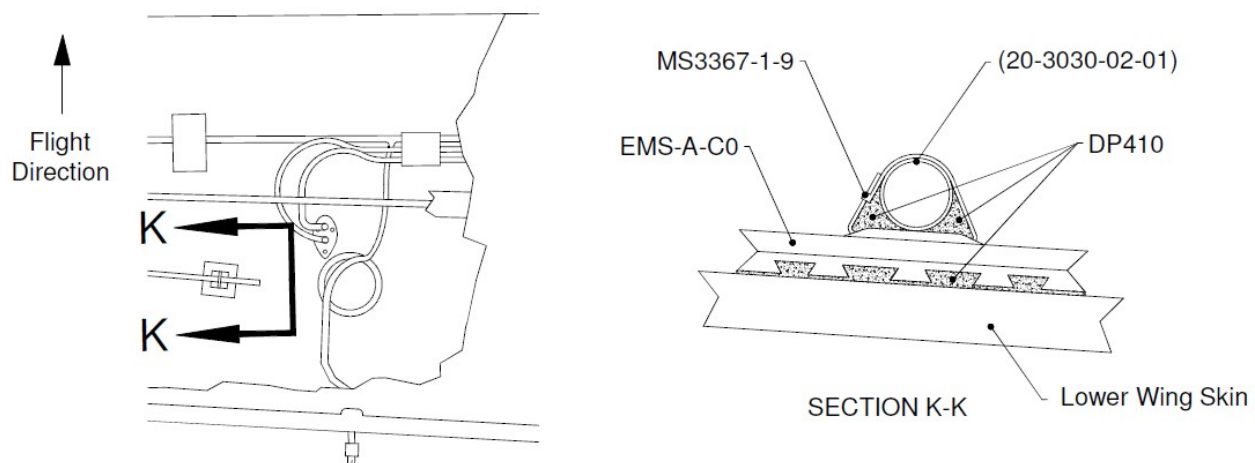


Figure 40: Bonding of Pitot Wire Conduit to Lower Wing Skin

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- 13.1.13** Bond a Cable Tie Base (p/n: EMS-A-C0) to the Spar Web using DP410 adhesive, as shown in Section L-L of Figure 41. The location of the bond along the wing spar is approximate.

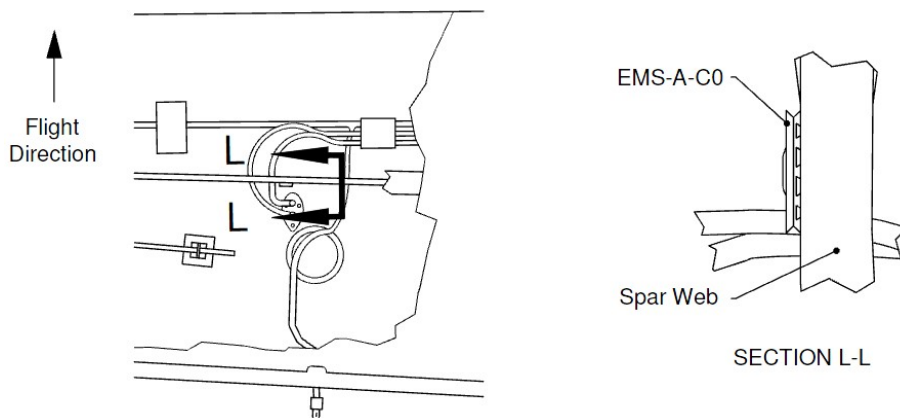


Figure 41: Bonding of Cable Tie Base to Wing Spar

- 13.1.14** Clean the work area around the Access Hole and the Landing/Taxi Light Cut-Out and inspect for foreign objects.

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13.2 Wiring Installation for Wing

NOTE: Section 13.2 will not need to be performed if MOD 0343 or OÄM 22-0122 is installed.

13.2.1 Follow the instructions below to assemble the wire connection for the wing to the fuselage, as shown in Figure 42.

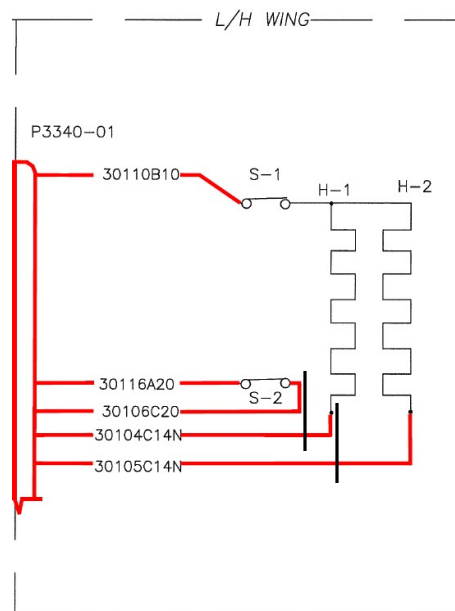


Figure 42: Wing Schematic highlighting Connection with Fuselage

13.2.1.1 Prepare the High Thermal Limit Switch wire according to Figure 43.

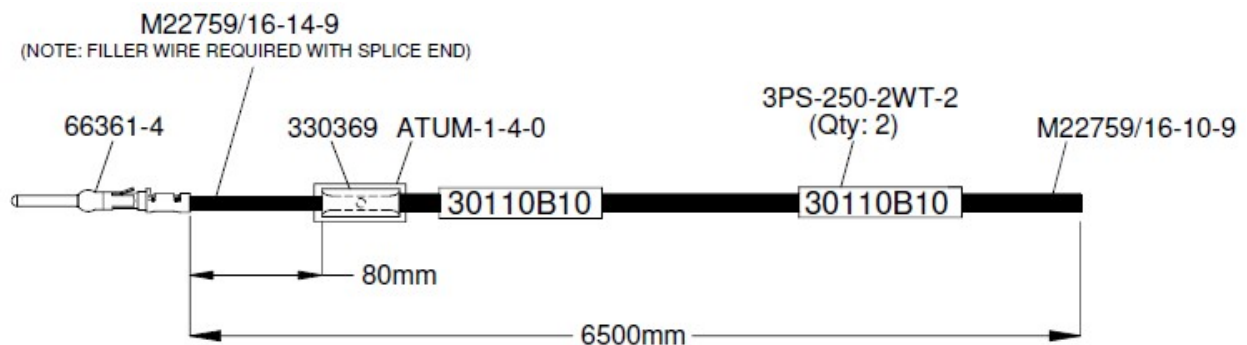


Figure 43: Preparation of High Thermal Limit Switch Wire

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13.2.1.2 Prepare the Low Thermal Limit Switch wires according to Figure 44.

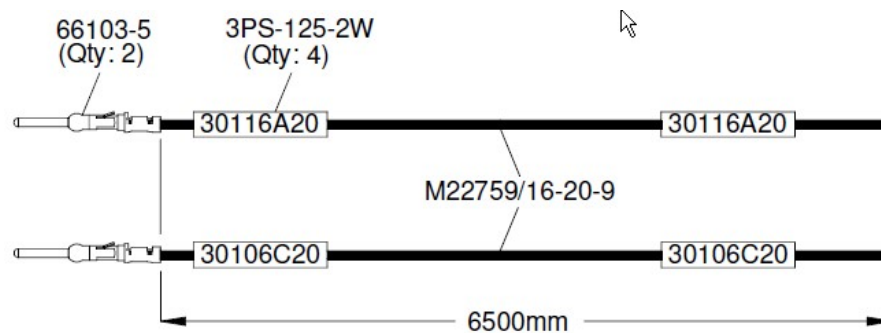


Figure 44: Preparation of Low Thermal Limit Switch Wires

13.2.1.3 Prepare the ground wires according to Figure 45.

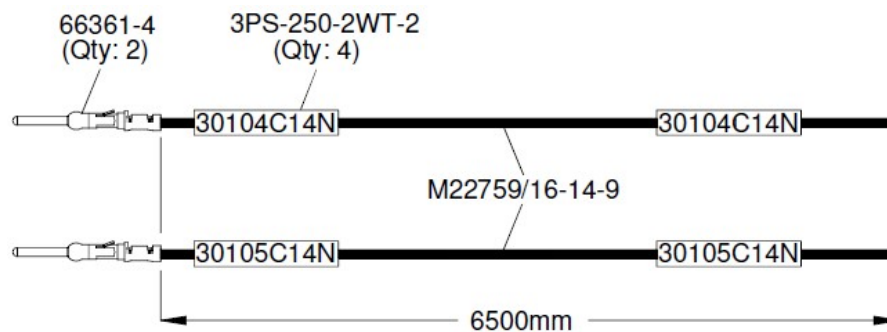


Figure 45: Preparation of Ground Wires

13.2.1.4 Install the wires into the existing wing P3340-01 Plug, located at the left hand wing root, according to Figure 46.

P3340-01	
PIN	WIRE
21	30110B10
20	30116A20
19	30106C20
16	30104C14N
22	30105C14N

Figure 46: Wing Plug for Fuselage Receptacle

13.2.2 Connect the wing P3340-01 Plug with the fuselage J3340-01 Receptacle and secure the connection with supplied cable ties (p/n: MS3367-5-9).

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- 13.2.3** Route the prepared wires through the wing conduit and along the existing wire harness for the wing. Pull the wires out of the hole in the wing conduit at the Landing/Taxi Light Cut-Out and route them through the Pitot Heat Conduit towards the Pitot Probe Opening, as shown in Figure 47.

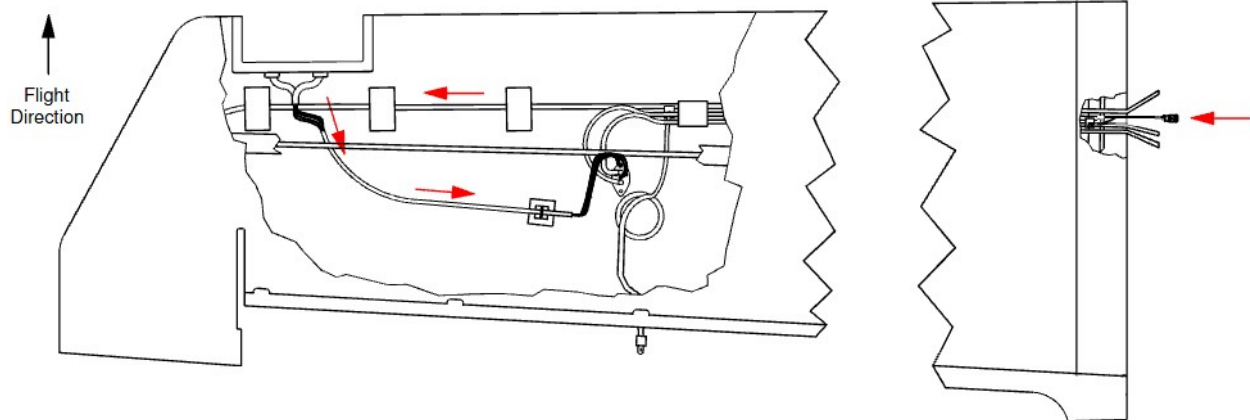


Figure 47: Routing of Wires in Wing

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13.3 Pitot Probe Wiring and Installation

NOTE: If MOD 0343 or OÄM 22-0122 is installed, remove cap, and stow from wires at the pitot mast.

- 13.3.1** Assemble the Low Thermal Limit Switch (p/n: M24236/1-0280) and the Bracket Assembly (p/n: 22-3030-60-00) according to Figure 48. Apply Heat Transfer Compound (p/n: HTCP-20S) between the contact surfaces of the Low Thermal Limit Switch and the Bracket Assembly. Wipe away any excess.

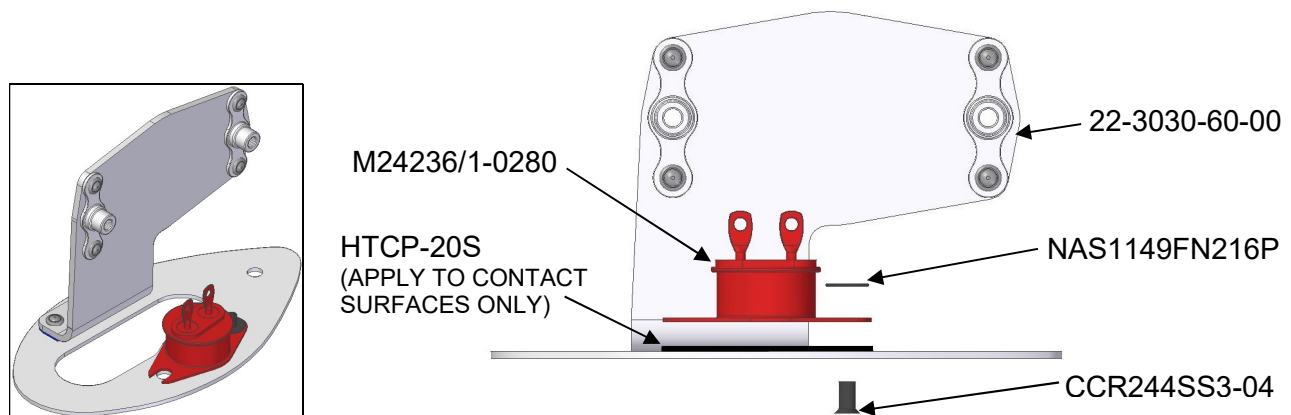
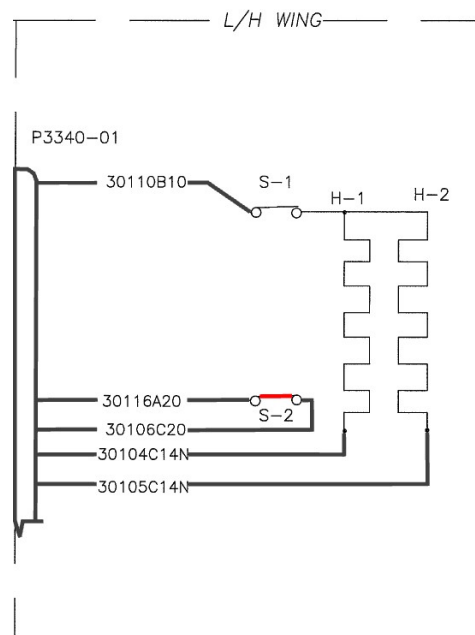


Figure 48: Assembly of Low Thermal Limit Switch and Bracket Assembly

- 13.3.2** Follow the instructions below to install the Low Thermal Limit Switch Assembly, as shown in Figure 49.



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Figure 49: Wing Schematic highlighting Low Thermal Limit Switch Installation

- 13.3.2.1 Place the Inner Phenolic Gasket (p/n: 22-3030-00-11) inside the wing and position the Outer Phenolic Gasket (p/n: 22-3030-00-09) and the Bracket Assembly according to Figure 50.

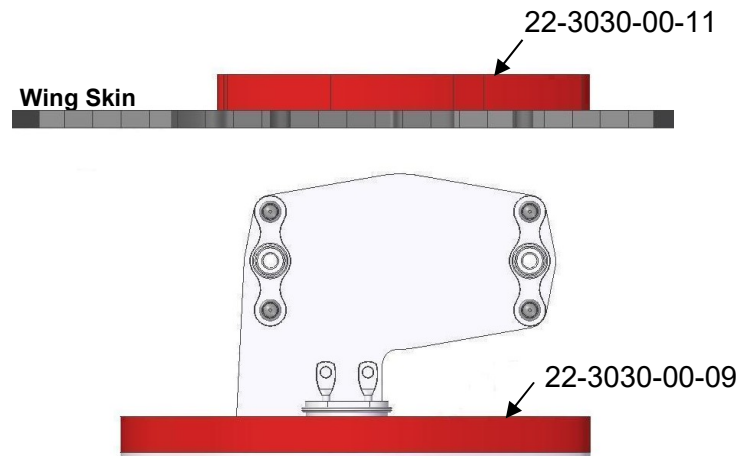


Figure 50: Positioning for Low Thermal Limit Switch Installation

- 13.3.2.2 Temporarily route wires 30116A20 and 30106C20 through the Pitot Probe Opening, as seen in Figure 51. Trim wires to length according to aircraft standards and practices.
- 13.3.2.3 Solder wire 30116A20 to either lead of the Low Thermal Limit Switch. Use supplied heatshrink (p/n: ATUM-1-8-0) for strain relief.

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13.3.2.4 Solder wire 30106C20 to the remaining lead of the Low Thermal Limit Switch. Use supplied heatshrink (p/n: ATUM-1-8-0) for strain relief.

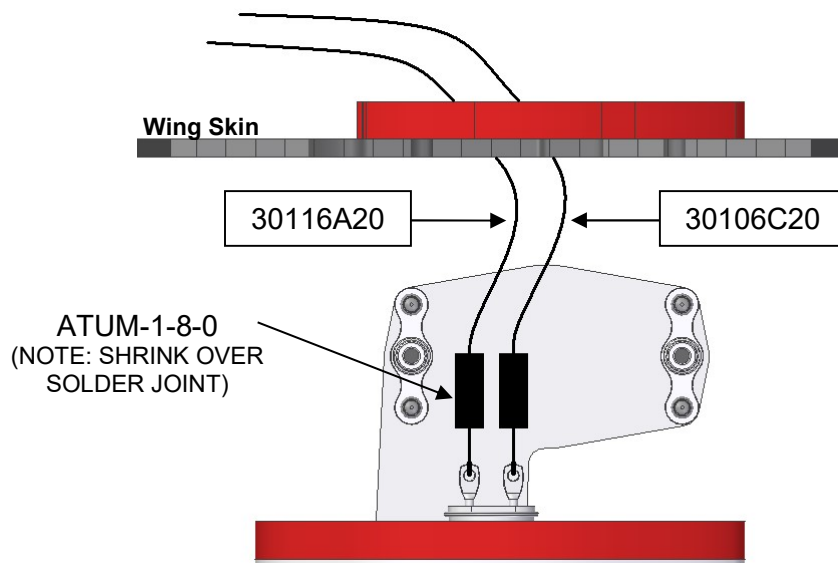


Figure 51: Low Thermal Limit Switch Solder Connections

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- 13.3.3** Install the Heated Pitot Static Probe Assembly (p/n: 22-3030-01-00) on the wing according to Figure 52 (wires not shown for clarity) and AMM # DA201-C1, Chapter 34-10. Apply a thin layer of Thermal Conductive paste, a maximum $\frac{1}{2}$ mm thickness, (p/n: HTCP-20S) between the contact surfaces of the Probe Assembly and the Bracket Assembly. Wipe away any excess paste after assembled.

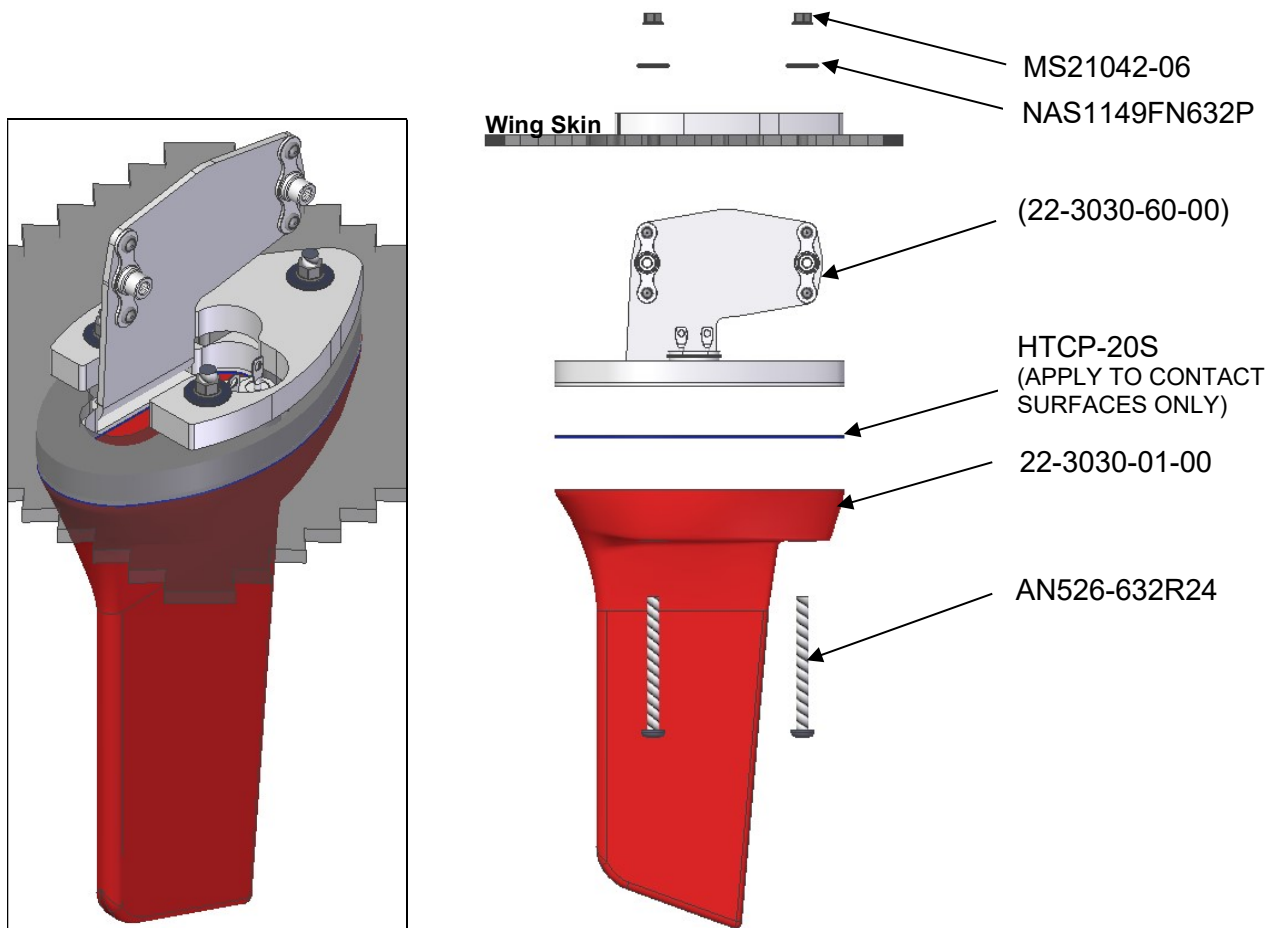


Figure 52: Heated Pitot Static Probe Installation

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- 13.3.4** Follow the instructions below to install the wiring for the High Thermal Limit Switch (p/n: 4391S14-209), as shown in Figure 53.

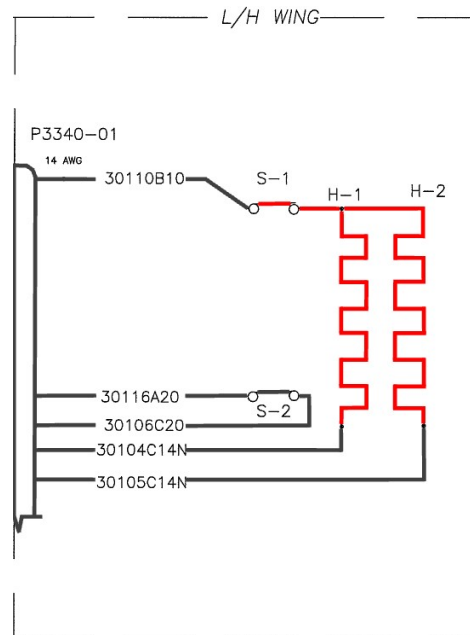


Figure 53: Wing Schematic highlighting High Thermal Limit Switch Wiring Installation

13.3.4.1 Trim wires 30110B10, 30104C14N and 30105C14N to length according to aircraft standards and practices.

13.3.4.2 Splice together wire 30110B10 with either lead of the High Thermal Limit Switch according to Figure 54.

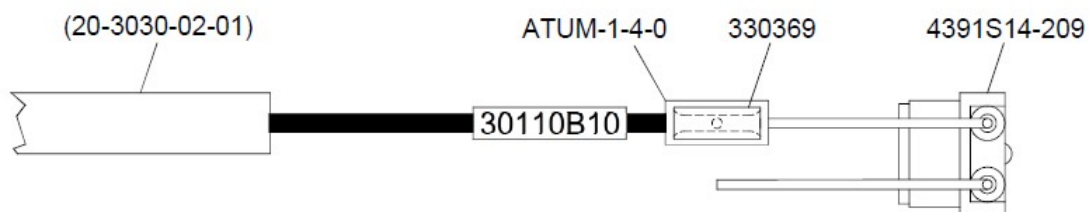


Figure 54: High Thermal Limit Switch Wire Installation

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- 13.3.4.3 Splice together the 2 heater element leads from the Heated Pitot Static Probe Assembly with the remaining lead of the High Thermal Limit Switch, according to Figure 55. Use the splice provided with the Heated Pitot Static Probe Assembly.

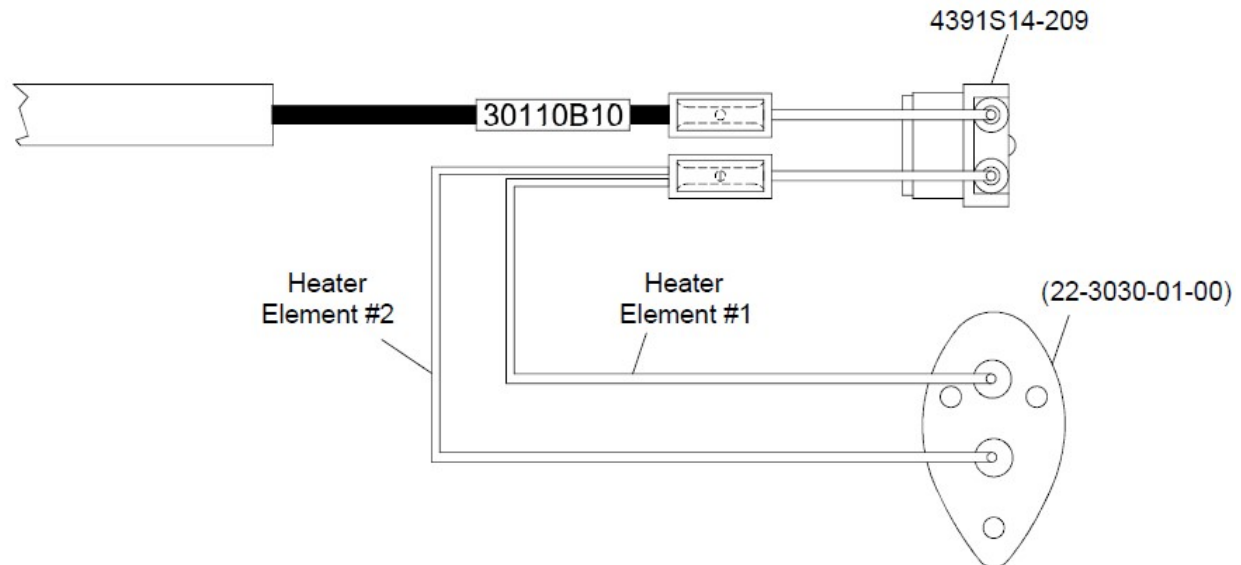


Figure 55: High Thermal Limit Switch and Pitot Probe Wire Installation

- 13.3.5 Splice together wire 30104C14N with the ground lead of one of the heater elements, from the Heated Pitot Static Probe Assembly. Use the splice provided with the Heated Pitot Static Probe Assembly. See Figure 56.

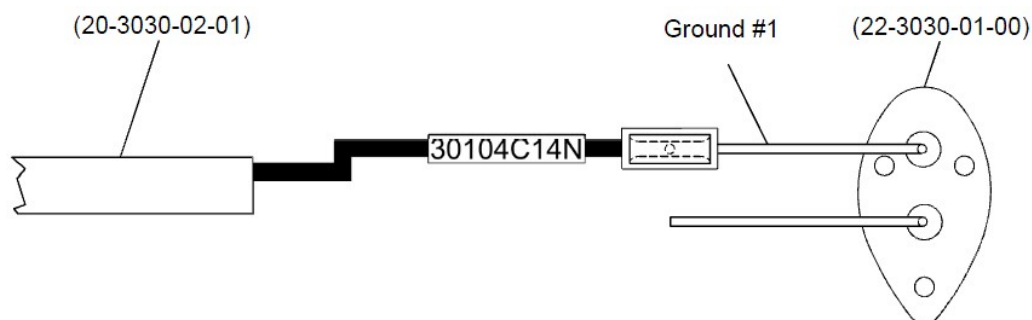


Figure 56: Heated Pitot Static Probe Assembly Ground Wire #1 Installation

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- 13.3.6** Splice together wire 30105C14N with the ground lead of the remaining heater element, from the Heated Pitot Static Probe Assembly. Use the splice provided with the Heated Pitot Static Probe Assembly. See Figure 57.

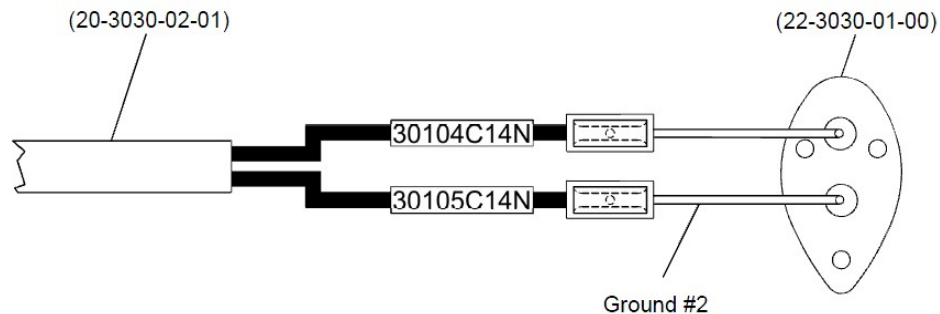


Figure 57: Heated Pitot Static Probe Assembly Ground Wire #2 Installation

- 13.3.7** Install the High Thermal Limit switch to the already installed Bracket Assembly (22-3030-60-00) according to Figure 58. Apply a thin layer of Thermal Conductive paste, a maximum $\frac{1}{2}$ mm thickness, between switch and bracket. Wipe away any excess paste (p/n: HTCP-20S) after assembled.

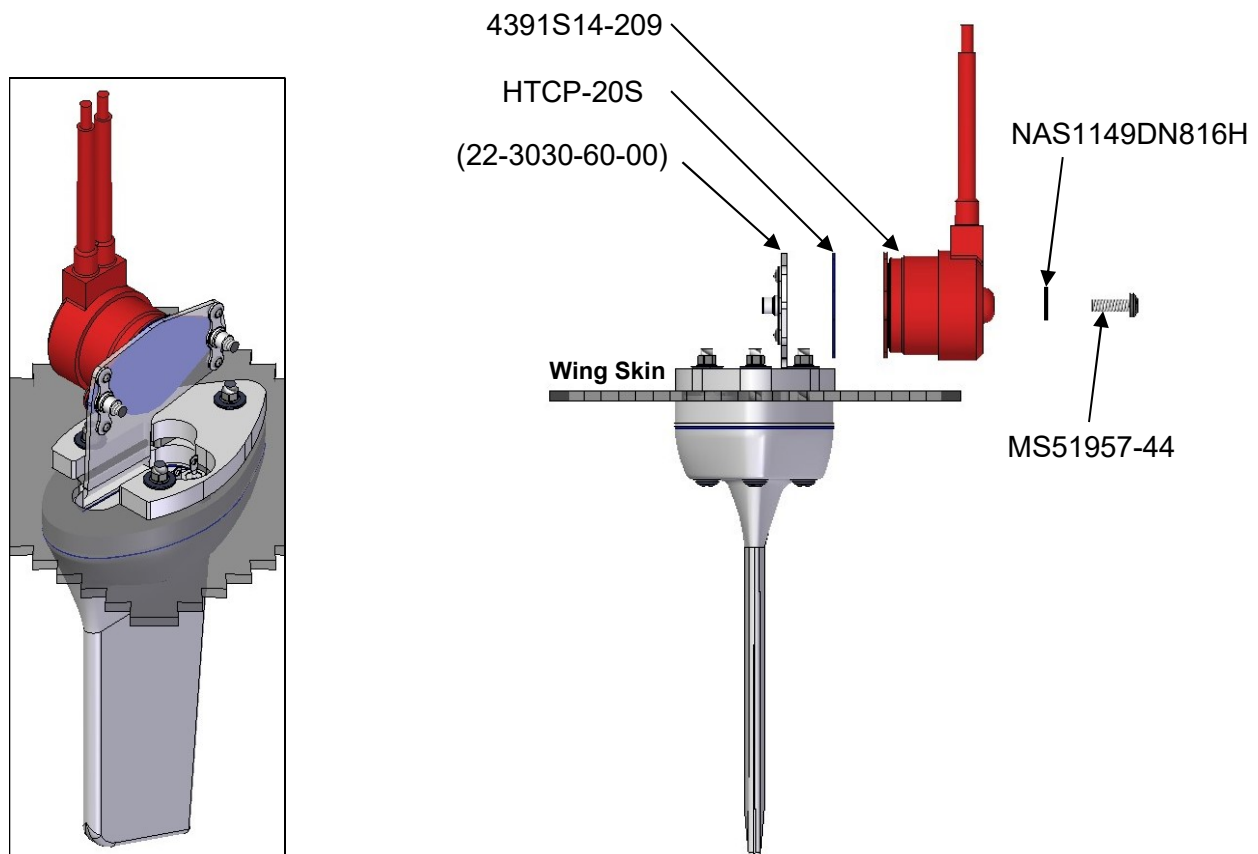


Figure 58: High Thermal Limit Switch Installation

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- 13.3.8** Secure wires 30110B10, 30116A20, 30106C20, 30104C14N and 30105C14N to the Cable Tie Base (p/n: EMS-A-CO) on the Wing Spar using supplied cable ties (p/n: MS3367-5-9).
- 13.3.9** Secure any remaining slack wires using acceptable aircraft standards and practices.
- 13.3.10** Clean the work area around the Pitot Probe Opening and the Access Hole and inspect for foreign objects.

14. Accomplishment Instructions: CLOSING

14.1 Closing

- 14.1.1** Inspect each work area for FOD.
- 14.1.2** Close the outboard Access Panel on the underside of the wing.
- 14.1.3** Install the Landing Light and Taxi Light according to AMM # DA201-C1, Chapter 33-40, if required.
- 14.1.4** Install the LH Wing according to AMM # DA201-C1, Chapter 57-10, if required.
- 14.1.5** Install the Pilot Seat according to AMM # DA201-C1, Chapter 25-10.
- 14.1.6** Install the Instrument Panel Cover according to AMM # DA201-C1, Chapter 25-10.
- 14.1.7** Connect the Aircraft Battery according to AMM # DA201-C1, Chapter 24-31.

14.2 Operational Check

- 14.2.1** Follow the instructions below to perform an Operational Check on the Heated Pitot Static Probe.
 - 14.2.1.1 Close the BATTERY Circuit Breaker and power the Instrument Panel.
 - 14.2.1.2 Set the PITOT HEAT Rocker Switch to "ON" and ensure the green LED light on the PITOT HEAT Rocker Switch illuminates.
 - 14.2.1.3 Check the Heated Pitot Static Probe Assembly for a temperature increase.

WARNING: THE PITOT STATIC PROBE CAN GET VERY HOT AND CAUSE BURNS. TAKE CAUTION TO PREVENT INJURY.
 - 14.2.1.4 Set the PITOT HEAT Rocker Switch to "OFF" after verifying a temperature increase. Ensure the green LED light on the PITOT HEAT Rocker Switch disappears.
 - 14.2.1.5 If Pitot/Static lines are disconnected throughout this installation a leak check must be performed. Refer to AMM # DA201-C1, Chapter 34-10.
 - 14.2.1.6 Turn aircraft power off.
- 14.2.2** Make a logbook entry that this Service Bulletin has been incorporated.

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15. Weight and Balance: An aircraft weight and balance must be performed and recorded upon completion of this service bulletin.

16. Availability: Contact Diamond Aircraft Industries Inc.

17. Electrical Load Data:

Description	Current (A)
Pitot Heat System	13.1

18. Credit: None.

To obtain satisfactory results, procedures specified in this service bulletin must be accomplished in accordance with accepted methods and current government regulations. Diamond Aircraft Industries Inc. cannot be responsible for the quality of work performed in accomplishing the requirements of this service bulletin. Diamond Aircraft reserves the right to void continued warranty coverage in the area affected by this service bulletin if it is not incorporated.

If you no longer own the aircraft to which this service bulletin applies, please forward it to the current owner and send the name of the current owner to Diamond Aircraft Industries Inc. at the address below.

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Appendix A – Scaled Template of Wing Cut-Out

